

*Aeronautical Engineering Section*

June 15, 1929

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# AVIATION

*The Oldest American Aeronautical Magazine*



*SPECIAL FEATURES*

*Air Transport* PROGRESS IN THE UNITED STATES

*Aircraft Control* OF LIGHTING FOR  
EMERGENCY LANDING FIELDS

AERONAUTICAL *Engineering* SECTION

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[illegible]

...help if you would enjoy the same  
...help.

AA

**The ALLIANCE AIRCRAFT CORPORATION, Alliance, O.**

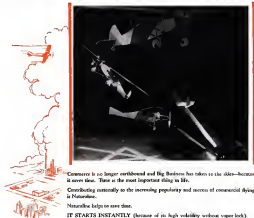
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## Dimensions

Span upper wing . . . 35' 4"  
Span lower wing . . . 31' 4"  
Total wing area . . . 322 sq. feet  
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Available cargo space . . . 30 cu. feet



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DEALER: "Exactly—now that you're going to be operating in the Wisconsin lake region, this biplane amphibian just fits your requirements."

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DEALER: "Yes—and it's a mighty important step. It means that you can get with your plane the same sort of service you have come to expect with your automobile."

PASSENGER: "Sounds good to me. I'd like to fly that biplane if you could arrange it."

DEALER: "Certainly. There's Captain Ervin now, let's see our New England manager. I'll introduce you and we can arrange a demonstration immediately."

PASSENGER: "First! You fellows certainly work in close harmony."

DEALER: "You bet we do, and it means a lot to all of us."

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## AVIATION

THE OLDEST AMERICAN AERONAUTICAL MAGAZINE

June 15, 1929

Volume 11 Number 1



## Bigger and Better Airports

**A**N AIRPORT can be too large. The larger it is the better chance a pilot has of his engine quits on the take off and the better are his chances of finding it in bad weather. Also it can be divided into take off and landing areas, and certain parts can be reserved for instruction and others for transport planes.

Large cities can afford to build and maintain large airports even in spite of the high cost of land because of the intensity of traffic. The big passenger airlines are in reality being designed around these large airports. They could not be operated from many of the fields which existed a few years ago or which now exist outside of the larger towns. Their landing speed and the momentum of their weight is too great to allow them to operate safely out of small rough fields. The airplanes of tomorrow, as the ones known of today, will probably only be able to operate out of certain large ports.

The tendency of the aircraft designers to increase landing speeds as fast as airports are improved is perhaps deplorable, but it is also inevitable. It presents a serious problem for the small towns which happen to be along the line of major air traffic. Such a town can not afford to purchase, put in the equipment, and maintain a field of sufficient size to handle our big passenger planes. It would seem as if the individual states would have come to the aid of the municipalities in regard to the major airports of our transcontinental routes.



## Unified Activity

**W**ITH the rapid growth of the aviation industry there has come a most serious growth of organizations whose purpose is to promote flying in one way or another. We have organizations of pilots—mechanics, manufacturers, dealers, distributors, the general public, and many subdivisions of each, all seeking to advance the cause of aviation.

Too many organizations trying to do the same thing can overlap in their activities, blind the aviation industry

of time and money and greatly disprove the energy being expended for the advancement of aviation. It would be well to remember that, "In unity there is strength," and though it may hurt certain groups or individuals, the various groups and causes must be consolidated wherever possible in order to unify the activities being conducted, to conserve the time, money, and energy of the valuable men who are giving their efforts to this work and to present a united front to the many grave problems with which the aviation industry is now faced.



## Official Designs

**T**HE DEPARTMENT of Commerce has decreed that all planes shall come out of a shop in three quarters of a turn after having made six turns. This is undoubtedly desirable, but it is doubtful whether, with our present knowledge of design, this really adds to the safety of flying. Regrettable as it may be, we have not yet designed planes which will not spin or stall. It is also unfortunate, but apparently true, that a plane which is hard to get into a spin is also hard to get out of a spin. Conversely, a plane which is easy to get out of a spin is also easy to get into a spin.

According to available figures, the majority of airplane accidents start at altitudes of less than three hundred feet, and at this height it is very difficult to get out of a well developed stall or spin. Designers are doing their best to develop planes that will only spin under unusual circumstances and which will come out of their spins easily.

This safety measure which has been adopted by the Department is distinctly open to controversy. What is even more serious, is that it is not more step toward the complete elimination of aeronautical engineering by the officials at Washington. It is easier to pass safety regulations affecting design than it is to repeal them, and it will not be long before the civil handbook controls commercial airplane design as thoroughly as the Army and Navy handbooks control military design. Such a condition would be our end if deplorable.

# Air Transport

## PROGRESS IN THE UNITED STATES

**W**ITHOUT DOUBT, the United States is out-  
racing upon the greatest era of air transport  
which the world has yet witnessed. The past 30 years  
dedicated to enter in the definite column, in a larger  
way than ever, the results of many years of pioneering,  
pioneering effort and accomplishment. Ample capital  
is available, the aircraft manufacturers are developing  
and producing without limitation airways are being es-  
tablished, and nothing stands in the way of a  
full realization of air transportation in all its phases and  
with all its possibilities.

Since the World War, when a large quantity of surplus military aircraft was released for sale, commercial air transport has been in the making. First came the beginning of the air mail service with many pilots flying

the route between New York City and Washington, D.C., the establishment of the Transcontinental Airway  
Route, an air mail service from coast to coast. Later, the  
opening of the route, the addition of other routes, and  
now the great 20,000 mi. network of aerial highways  
with their lighting equipment, transatlantic fields, at-  
tendants, radio and weather service, are all a matter of  
history, and air mail has become an essential part of our  
daily lives.

Only in the last few years, almost in recent months,  
have the commercial air transport companies named their  
airlines seriously to the transportation of passengers.  
There have been pioneer lines to be sure, the majority of  
these operating successfully and furnishing excellent  
service, but the current year marks the real first step—

the enlargement of this service to an  
extent that will cause us to take it for  
granted and to depend upon it, even  
more than we have come to accept and  
rely on the air mail.

**W**E HAVE ALREADY witnessed the  
transportation of several important  
services, both national and inter-  
national. They have demonstrated  
their need. We are to have at least  
two additional transcontinental lines  
devoted primarily to passenger service,  
and practically all air mail contractors  
are making plans to include it, with  
larger and more powerful planes to  
supplement those which have been pro-  
viding the routes through all sorts of  
weather and throughout the twenty-  
four hours of the day. Passengers  
will ride comfortably in easy chairs  
and berths, their meals attended by  
uniformed stewards, and traveling by  
air will take on the comfort of a  
Pullman journey.

Within the United States, there  
exist all the fundamentals needed to  
foster successful air transport lines  
and service to any extent for which

*An Exceptionally Interesting Description of  
Present Conditions and of How We  
Have Profited from the Experiences  
of European Airway Operators*

By MAJ. CLARENCE M. YOUNG  
Director of Aeronautics, U. S. Department of Commerce



there could possibly  
be a demand. Our  
present transport  
planes measure up to  
the best to be  
found elsewhere.  
They compare fa-  
vorably with those  
in use by the im-  
mense airlines of  
Europe, such as the  
Imperial Airways  
of Great Britain,  
the Luft Hansa of  
Germany, and the  
K. L. M. of Hol-  
land. And there is  
no question but  
what the aircraft  
and engine manu-  
facturers of the  
United States, who  
have made such  
transatlantic flights  
in the last few  
years as the de-  
mand for new and  
more efficient air-  
craft has developed,  
will keep us in the  
van of other na-  
tions in the produc-  
tion of large, pow-  
erful air liners that  
will be very safe and  
extremely comfortable as well.

We possess already a vast network of properly  
equipped airways which follow, in the main, the trade  
routes of former eras of transportation. These airways  
are constantly being added to, and the new lines will  
be practically every important industrial center in the  
United States linked up in this rapidly developing aerial  
highway system.

Due to the lightning and enterprise of both man-  
agement and individuals, there are being developed in the  
United States some of the finest airports to be  
found in the world. And when it is realized  
that the major portion of this work has been  
undertaken within the last three years, one  
cannot help but be impressed with the transi-  
tory possibilities, immediately ahead, and the  
extent to which we already rely, perhaps un-  
knowingly, upon air transportation.

During the last three years privately own-  
ed and companies have developed scheduled  
air transport services carrying 50,000 air daily  
—twice the distance around the earth at its great-  
est circumference. Schedules are maintained  
with precision throughout the 24 hr. of each  
day, over mountain ranges and through all sorts  
of weather. During this same period personnel  
has been trained, airways have been established,  
radio and weather services organized, extremely  
valuable experience has been gained, and it all  
furnishes the promise for added transportation  
service that will equal or exceed any which  
can be dreamed of.

In giving consideration to the existing air  
transportation services in the United States,  
from any point of view, it is necessary to be  
mindful of the fact that they have been de-  
veloped without government subsidy, or direct  
aid of any kind, such as many European lines  
enjoy. Sometimes one is apt to confuse air  
mail contracts with subsidies in some form  
or other, but such is not the case. To be sure,  
the transportation of air mail by private enter-  
prise has been the important factor in building  
up existing scheduled services, but air mail contracts are  
awarded upon the basis of competitive bidding; and the  
holders thereof perform a definite service for any  
amount received. The daily service which benefits upon  
government aid is the establishment and maintenance of  
the airways, but this is similar to the light-house and naviga-  
tion service that has been rendered shipping for many  
years. It cannot, therefore, be considered in the light of  
a government aid peculiar to air transportation.

All things considered, air transportation in the United  
States is in a position to develop rapidly and to become  
one of the most important factors in our national life.



Chrysler Airport, which serves London, as seen from the gallery  
outside of the control office at the hotel.

States is building upon its own premise with its own initiative, and will advance to any stage where a service can be rendered—and the limitations are comparatively few.

The question has frequently been asked—what have we to learn from the extensive experience gained in the transportation of passengers in various European op-

erations over a period of approximately two years? It cannot be answered in an off-hand manner, because of some important differences which become manifest when comparisons are made.

A fundamental which must exist in any successful air transportation system, whether it be in the United States, Europe, or elsewhere, is made up of a combination of three things: reliability, speed, and comfort. This involves practically all elements entering into such a system—aircraft, pilots, ground personnel and equipment, airways, schedule arrangements, rules and weather services, etc. No substantial variation is permissible. Any differences which may exist are, therefore, limited to such things as the extent of the service, the schedules measured the length of the routes, and to the details and refinements which are developed as experience is gained.

A number of European lines are operating excellent services. Some of these involve difficult conditions, such as winter crossings, inaccurate houring, fog, etc., and one cannot help but pay his respects to the success with which their operations have been conducted. Their routes, as a rule however, are comparatively short, a circumstance which simplifies schedule maintenance and operation to a considerable extent. For example, the route between London and Paris is but two and one-half hours; between London and Brussels a trifle less, with an additional hour and ten or fifteen minutes between Brussels and Cologne; London and Amsterdam is relatively the same, while Paris and Cologne are approximately three hours apart, and Cologne and Berlin about three and one-half hours. A somewhat similar condition exists from Paris south to Marseilles, etc., and from the actual operating point of view, a favorable set-up has been developed.

The routes and schedules in the United States are substantially different. A transcontinental service involves an airtime distance of approximately 2,700 mi. It crosses three mountain ranges, and calls for both night and day flying, and requires aircraft properly adapted to such a service.

EVERY SUCCESSFUL solution exists in earth and south operation, or in practically any direction where through services are contemplated—and it is the through, long-haul service which offers the greater economies in the matter of time, and justifies the extra differential which the users of transportation are willing to pay. Accordingly, there are many locations in the United States where conditions of terrain, or the absence of other transportation, make air transportation only comparatively short contacts advantageous and profitable, but because of the vast area of the country, lengthy schedules need be contemplated when

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consideration is given to air transportation as a competitive national service. The difference in this respect between air transport operations in the United States and in Europe is at once quite apparent.

THE ASPECTS of the routes and the nature and extent of the services also reflect itself in the amount of equipment needed on airways. Radio communication and weather services become somewhat more complicated, intermediate fields, night lighting equipment such as beacon lanterns, markers, obstruction lights, flood lights in terminal fields, etc., all enter the picture and complicate the air transport operators in the United States with substantially different conditions than exist elsewhere, and with which they undoubtedly have had the greatest amount of experience.

To affect any disadvantages which may seem to present themselves in connection with longer routes, more ground equipment, etc., (and they are merely differences rather than disadvantages) is the fact that the services operating wholly within the United States are, or will not be concerned with the more or less varying lengths of passport customs, etc., caused by the many border crossings in Europe. Many of the air journeys, domestic though they are, last at two or three hours' duration, put one in entirely different countries with different languages, different customs, and involve the total examination of passports and luggage, and the formalities of immigration, public health, etc. Here one can be almost 3,000 mi. and still be in the same country and never see nor hear of a passport or customs official. It all reflects distinct advantages in the general scheme of things and will react most favorably in our air transportation system.

THE ECONOMY is by no means an attempt to indicate profit for air transport operations in this country except that by the long operating experience of the European lines. As a matter of fact, a number of the representatives of such organizations have visited Europe at various times for the purpose of studying their methods, and undoubtedly many things have been put into effect here as a result. At least one important phase are the details and refinements—the comparatively minor things which either irritate or greatly ease the situation, and enhance the pleasure for their attitude toward the system as a whole. The manner in which they are transported to and from the airport, the facility with which luggage is handled for them, the ease with which they are transferred to and from the aircraft, their comfort in the aircraft while en route, etc., furnish examples of the same referred to. Such details are dated into the present only as a result of experience—an acquired knowledge of the needs and desires of the persons con-

cerned. This may appear relatively unimportant—the actual transfer of passengers from one location to another may seem to dominate all other phases. It does, of course, but not in the mind of the passenger. His other duty, or will, rule that, for granted, but he will not overlook the fact that he was obliged to transfer his own luggage in a taxi to the airport, had his own way to the aircraft via the propeller belt, locate his own seat reservations, arrange himself as to how he was to board his location he is passing over, whether the aircraft is on schedule, where a market is when the aircraft takes the cable chair, why the belt contraption on his seat, and to worry about the aircraft hovering around in rough air without knowing that it is a normal condition in the particular locality. These and many other minor things are of slight consequence to the operators' concern, but they very definitely have their mark on the passenger—and after all he is the source of revenue and must be the one who is satisfied. The air transport organizations, therefore, would seriously do well to take such items into major consideration and put them into effect in the beginning rather than to later attempt to overcome any prejudices which their shrewd will surely create. Most of the European lines have nearly gone this phase of it a great amount of thought and development. And this is one reason why the European air passenger lines are so highly regarded by the layman today.

When the United States a thorough knowledge of the fundamentals of air transportation exists, extensive experience with the operating conditions has been gained, services have been established, equipment and personnel are available, and the users of transportation of any kind will soon find themselves confronted with air services in comprehensive and satisfactory that they cannot help but use them and eventually take them for granted in the same manner they do the railroad, the steamship, and the automobile.



Above: Major Young standing in front of the "Kaiserhof" airport, showing the location to which he made the run of 10 miles. Below: An air view of Flughafen Airport.



# THE GENERAL AIRPLANES

## "Aristocrat"

### CABIN MONOPLANE

**F**OLLOWING a long period of service tests in accord with its policy regarding new models, the General Airplanes Corporation, Buffalo, N. Y., has started production of the "Aristocrat" cabin monoplane. During the tests, which have been in progress since the first plane was completed in July, 1938, the craft has been flown by about 60 pilots and has had 90 hr. at 8,100 on its cross country service. The plane was recently uncovered and disassembled, inspected and rebuilt for further service tests.

The Aristocrat is a three place, externally braced type powered with the seven cylinder Warner "Saurik" radial air cooled engine which develops 110 hp. at 1,850 r.p.m. The plane has a wing span of 36 ft. 4 in., an overall length of 28 ft. 2 in. and an overall height of 7 ft. 7 in. The weight empty is 1,244 lb., the pay load 363 lb. and the gross weight, 2,110 lb.

In performance tests the Aristocrat developed a high speed (full load, at sea level and in level flight) of 165 m.p.h., high speed at 5,000 ft. of 165 mph, a cruising speed of 90 m.p.h. and a landing speed of 40 mph. The rate of climb at sea level is 650 ft. per minute and the plane climbs to 5,000 ft. in 5½ min. and 10,000 ft. in 24½ min. The service ceiling is 12,150 ft. and the absolute ceiling 14,900 ft.

Twenty-five airplanes were to be built during the first two months of production, twenty during the third month and twenty-five per month thereafter. The plane is manufactured under approved type certificate No. 117 and is conformant to the requirements of the Army, Navy and the I.C.A.N. One of these airplanes is being used by Commander Richard H. Byrd in the Aristocrat.

Wings are constructed of wood. The spars are of solid laminated spruce with additional hardwood facing strips at the point of strut attachments. The ribs are constructed of aluminum. The fuselage is constructed of spruce and reinforced with plywood. Standard Air Service type down rods, one set at the top of the spar and one set at the bottom of the spar, are employed as the method of internal drift bracing. The ribs are attached to the spars by means of hardwood angle blocks, secured and nailed on each side. The drag ribs are a specially constructed box type with straight spruce members between the spars, top and bottom. There is a curved spruce top strip at the top to form the wing contour, and plywood reinforcement on either side of the spruce members.

The wing curve employed is the General Airplanes

Corporation G.A.C. No. 300 and is the high lift type of curve, giving ample depth for an economical design of wing spars. The aluminum are of the exterior edge built-up type, extending the full length. The aluminum control is entirely made of the wing and the aluminum bearings, as well as the bearings on the operating mechanism, are of the built bearing type. A 20 gal. gasoline tank is carried in each wing root to the fuselage and the drag bracket at this point is piped out the outer surface of the wing. The rib spring, outside of the propeller attachment, is 18 in., with extra nose ribs between the main ribs.

The contour of the exterior edge of the wing is maintained by an aluminum alloy sheet, covering top and bottom as far back as the front spar. The wing is fastened to the fuselage structure by pin joint fittings with angle iron bolts, allowing plenty of bearing area. This is also the case at the point of attachment of the lift struts to the wing spars. The wing struts are of stream-line tubing with adjustable fittings at one end. Great care has been given the design of these fittings



Another view of the cable bracing system



Side view of the Aristocrat cabin monoplane. The landing gear is of the strut type.



both as to their attachment to the tubes and the allowance for large bearing area at the bolt connection.

The fuselage is of conventional design employing chrome molybdenum steel tubing throughout, with a detachable engine mount on which is mounted the oil tank and complete oiling system before the same is assembled to the fuselage proper. Hot oil made primer is pumped through the fuselage while the outside of the structure is protected with one coat of red oxide primer, on top of which is applied an aluminum varnish. The sides, top and bottom of the fuselage are made in the form of a Pratt truss. The lift struts from the wing are fastened to two sturdy flat sheet metal fittings completely surrounding the top of the fuselage. The fitting over the center section and top of the fuselage is of formed round aluminum tubing, welded and clamped to the steel structure, thus making a very light and rigid streamline fairer over the fuselage.

A rigid pressure of tubes under the pilot seat takes the landing gear shock absorber unit, which is composed of hard rubber compression disks. To these built-up

shock absorber units are fastened sturdy, yet light, aluminum alloy landing gear legs. These legs are provided just below the longnose with two large nose pins at the front and one of the leg, distributing the load well along the longnose. The legs continue as an overhang beam, tapering to the point of attachment of the axle which carries the wheel. The whole unit is nicely fitted with an aluminum alloy steering housing, made of which is carried the brake operating cable. The brake cables run over pulleys to the rubber pedals, and are operated by the pilot's feet, thus allowing the ball of the foot to be on the rubber pedal at all times. With this arrangement both brakes can be operated simultaneously or individually. This, in combination with the tail wheel arrangement, which operates through more than a 180 deg. angle, allows the pilot to take in almost any position with great ease. Tires 30 x 5 are standard equipment on the Aristocrat.

The type of landing gear used is quite simple and very durable. There are only two large axle bearings, which take all the wear. The tail wheel is an aluminum casting with roller bearings and a 10-in. x 3-in. pneumatic tire. It is mounted in a fork which is free to swing 360 deg. in an aluminum alloy casting hinged about a horizontal axis at the base of the tail post. Shock absorber cords are attached to the fork which prevent the latter from making a complete circle and which insure directional stability while taxiing.

A considerable amount of attention was paid to the design of the tail wheel and mechanism in order to eliminate wear on the working parts as much as possible. The tail post, at the point of attachment to the tail wheel gear, is perpendicular to the ground line with the tail down in order to give the desired cam effect. The rubber is biased to the tail post in two places and it is out to clear the tail wheel.

The tail surfaces are constructed of welded steel tubing. The stabilizer is braced by a strut from the lower end of the outboard end, in addition, a wire runs from the top of the stabilizer spar to the fin. The stabilizer is adjustable in the air by means of a self-locking square thread, operated by a drum and cable mechanism within reach of the pilot.

The flying controls are conventional and of the stick type. Great care was given to the design of the rick-type bearings in order to eliminate wear and slackness.

The elevators are operated from the stick control by means of cables under the floor. The ailerons are operated by means of cables from the stick by way of the forward lift strut to the trailing edge of the wing, where they give a bell crank which is fastened to the front spar. This bell crank operates the aileron-removing cable which passes through the neutral axis of both spars, thus keeping the entire aileron control inside the wing structure. Necessary inspection doors and windows are installed



A photograph showing one of the leading rear legs of the Armstrong. These legs are of direct construction and carry weight under the shock absorbers.

for proper check-up and inspection. The rubber is operated by means of extra flexible cable which passes from the rubber pulley over pulleys to the rubber lever.

The Warner "Scout" engine, developing 100 hp at 1,850 r.p.m., is standard equipment in the Armstrong. The structure is so designed as to accommodate an Eclipse hand starter. An exhaust collector ring is mounted on the forward part of the engine with a connection at the lower end passing through the carburetor air heater. A manually operated butterfly valve in this tube regulates the amount of hot air as that proper operating conditions exist in any temperature. A color heater can be supplied when desired.

By having the exhaust collector ring around the lower part of the cylinder and in front, a certain amount of heat is distributed over the crankcase and the lower part of the cylinder for cold weather flying. This collector attachment is so arranged that a manually operated shifter can also be installed to regulate the heat. In this way the Scouts can be flown down extreme high temperatures to extreme low temperatures and still maintain the desired temperature around the engine. A metal propeller is standard equipment on the Armstrong plane.



The cable heater with fuel and a portion of the front structure of the Armstrong showing the shock absorber springs of wheel strut tubes.

An inverted view of the fuel delivery system showing one of the delivery lines.

As previously stated the gasoline system consists of a 20 gal. tank in each wing seat to the fuselage. These tanks have a half-gal. pump in the bottom into which is attached a water drain and a finger strainer in accordance with Air Corps standard practice. The line from the bottom of the fuselage to a three-way valve connection, whereby gas can be taken from either or both tanks or both tanks can be balanced. From this three-way valve connection a line runs to the lowest point in the gasoline system at which point a C-I Air Service type strainer is installed. A line with flexible connections at the strainer and at the carburetor completes the system. Great attention and careful consideration have been given the gasoline system to make sure that the engine is properly fed during all maneuvers of flight, and at no point in the line enters the cabin. The gasoline tanks are usually thick aluminum sheet with welded joints and are extremely rigid but quite light in weight.

THE OIL TANK is of 4 gal. total capacity, made of the same material as the gas tank and is mounted between the engine and the fire wall and is detachable with the engine and the engine mount. A one-piece fire wall, which is attached to the main fuselage structure before the detachable engine mount is installed, gives the required rigidity so that no oil or gas fumes can penetrate into the cabin.

Large, well upholstered seats and hinged glass sliding windows along the entire cabin give the passengers all that can possibly be required in the way of comfort, light and good vision. The hinged glass windows just back of the pilot and on each side of the cockpit slide back to allow the pilot good vision forward and down to land weather flying.

The Warner engine is so mounted on the front of the plane as to allow the pilot good vision over the top of the engine as well as out of the side windows of the plane.

The pilot sits alone in the front part of the cabin. The



The cable heater with fuel and a portion of the front structure of the Armstrong showing the shock absorber springs of wheel strut tubes.

An inverted view of the fuel delivery system showing one of the delivery lines.

fuelage, while giving plenty of room at this point, is narrow enough to ensure good visibility out of both sides. Where dual control is required, arrangements are made to carry a second control stick and a second set of pedals to be operated from the rear seat of the Armstrong.

The cabin of the Armstrong is upholstered in two-tone effect, the color of which blends with the outside scheme of the plane. An instrument panel, containing altimeter, tachometer, oil pressure gage and oil thermometer, is

Photograph showing the elevator control rods and the fuselage structure of the Armstrong's main fuselage after it has been removed from the assembly line.



A view from the rear showing the "tail-wheel" effect.

located on the dashboard. This panel together with the air speed indicator, engine switch, gauges, flying lights and other necessary equipment, gives a pleasing appearance to the front of the cockpit. A standard Air Corps type engine control unit is installed in the front cockpit to the left of the pilot.

The doors on either side of the cabin are supported by two sturdy hinges which are bolted to the structure. This prevents the doors from loosening up, a frequent occurrence when hinges are wood-screwed to a separate frame. The baggage compartment, at the rear of the cabin unit, is accessible from both the cabin and the outside of the plane.

The remarkable performance of the Armstrong is getting in and out of small fields and toward making it an ideal plane for cross-country flying. The Armstrong is easy to handle as it can be flown "hands off" under normal weather conditions.

The officers of the General Aircrafts Corporation are: Charles S. Remmen, president; Lellay A. Lustig, vice-president and treasurer; A. Francis Aron, vice-president in charge of operations; George A. Townsend, secretary and sales manager; Donald A. Boyles, assistant sales manager. These officers, together with the following, constitute the board of directors: Evan Haddock, H. J. Frey, Alfred Marches and Oscar Staffin. James H. Stevenson is assistant to the president and



general manager. A. B. South, factory superintendent; Fred Bennett, assistant factory superintendent; Leo Foster, chief inspector; and B. Z. Peck, field and service engineer.

The specifications furnished by the manufacturers are as follows:

Length overall	25 ft. 2 in.
Height overall	7 ft. 7 in.
Wing (wing section)	G. A. C. No. 300
Span	36 ft. 4 in.
Chord	6 ft.
Area of wings, total	194.5 sq. ft.
Area of elevator	19.46 sq. ft.
Area of horizontal stabilizer	12.24 sq. ft.
Area of elevator	10.97 sq. ft.
Area of fin	1.62 sq. ft.
Area of radiator	7.47 sq. ft.
Weight empty	1,248 lb.
Pay load (fully equipped)	365 lb.
Disposable load	662 lb.
Gross weight loaded	2,113 lb.
Power, push, hp. and r.p.m.	112 at 1,850
Power loading	10.5 lb. per sq. ft.
High speed (full load: sea level and 5 mi. high)	160 m.p.h.
High speed at 5,000 ft.	155 m.p.h.
Cruising speed (1,670 r.p.m.)	90 m.p.h.
Landing speed	40 m.p.h.
Take off run (full wt.)	305 ft.
Climb at sea level	650 ft. per min.
Climb at 5,000 ft.	94 min.
Climb at 10,000 ft.	25 min.
Service ceiling	12,150 ft.
Absolute ceiling	14,700 ft.
Gasoline consumption at cruising speed	64 gal. per hr.
Gasoline capacity (normal)	40 gal.
Range at cruising speed	340 mi.
Endurance at cruising speed	6 to 15 hr.
Cost of plane P. A. P., Buffalo, N. Y.	\$7,300

# Aircraft Control OF FOR EMERGENCY

**A**MONG the noticeable tendencies in the growth of aviation is the increased attention placed on suitable landing fields. With the acquisition of numerous landing fields, and the resulting necessity for lighting equipment, a new problem presents itself.

If the use of emergency fields is going to be extended, some means of control for floodlighting must be provided. It is not practical to keep these fields floodlighted during all hours of darkness. Illumination for landing fields calls for large amounts of electric power and expensive lamps, not counting the cost of the auxiliary equipment. The ordinary floodlighting projector using a 3-kw. lamp for its source of illumination and projecting approximately a million beam candlepower, costs fifty cents an hour for the power consumed, at current power rates and cost of the lighting bulb. Several such lights are required for adequate illumination of a runway. The future emergency field located on important air lanes between large cities cannot be illuminated during the hours of

darkness unless heavy air traffic could justify the expenditure.

Weathermen cannot be considered dependable, since in case of inclement weather they would not be able to direct a plane in distress sufficiently in advance to render assistance, also the element of human error enters in.

Radio, light and sound are the only means available for communication between aircraft and ground. Radio affords a medium of control that is present in somewhat expensive and the equipment is bulky and complicated. In addition, the limited wave channels at present available prevent extensive use of radio as a means of control. With the rapid advance being made in the art of radio, a solution of these difficulties may be reached to permit the security of the future to be equipped with a single fool-proof radiophone that can be used as a means of communication and control for emergency landings. While successful control by means of light has been

# LIGHTING LANDING FIELDS

By GUSTAVE E. HERBERLEIN  
Qualification Engineer,  
Hawthorne Electric and Manufacturing Company



A battery of 10 floodlights at the Yonkers Municipal Airport, New York, N. Y.

demonstrated by various experiments, the use of light sensitive devices has proven more or less limited. Most of the attempts at remote control have used a scheme whereby the light received by a photo-electric cell on the ground from an aircraft equipped with searchlights was used as the signaling medium. The chief limitation of this scheme is that the observer must locate his landing field with the plane searchlight before the control can be accomplished. Obviously this is an undesirable situation, since in case of fog or darkness the field cannot be seen. Since the device is dependent upon the intensity of the light beam, operation at a distance cannot be obtained. Further, the photo-electric apparatus must be disconnected during the daylight, another disadvantage.

**T**HESE means available for control purposes are sound. Here an advantage is present over that of light in that a sound wave is not usually directed. The pilot can locate a field in fog, or darkness or in daylight providing the signal from the plane can be picked up by an emergency field located within the radius of action of the sound device. In addition, if a sound device resonant at one particular frequency is used, operation can be had independent to a large extent of the volume of sound, permitting operation at a distance or at close range.

Some time ago a test was conducted between the night air mail flying from Cleveland, O., to Trenton Field, Pa., in which the airport lights were energized by a relay operating from the sound of the plane's engine. In this demonstration, the receiving sound relay was tuned to the particular engine note, and as soon as the plane came into the radius of action of the relay, the sound of the engine actuated the relay, turning on the flood lights. The sound relays could be operated at large distances by the noise of the plane's engine but this extreme sensitivity brought out one disadvantage. The noise of

engines on motor trucks or passenger cars, when the engine was speeded up, would recognize the relays and turn on the lights, even though the particular automobile was quite a distance from the field. Since the chief cause for forced landings is the case of engine failure, the use of the engine as a sound signal device is out of the question.

The sound signal device to be practicable must be of some special nature which will operate as long as the plane is in motion or aloft. A wind-driven relay will meet these requirements, for as long as the plane is in motion the relay can be operated. In addition, the relay offers other advantages. It is light-weight, simple, fool-proof and can be used for other purposes as well as being sound.

As the wind-driven relay starts to rotate from rest, the sound frequency varies from zero to some high pitch determined by the wind speed and the number of parts on the periphery. The fact that the relay operates between certain limits of frequency is particularly advantageous for use in control of the airport since the receiving relay can be set for one definite pitch and the relay must pass through that pitch as it gathers speed.

If a sound sensitive device is used as a relay, it must necessarily be one which will respond only to one particular signal and it must be sensitive and dependable. The airport lighting relay is built around that principle. The heart of the device is a selective reed relay and a Knowles "Grid-Glow" tube. The reed relay proper consists of a magnetically coated reed and controlled by an electromagnet. The reed is sensitive to only one pitch or frequency of the sound in the audible sound range. The Knowles tube is a neon gas tube which can be set in operation by an infinitely small amount of energy, but it will release an appreciable amount of energy. The tube can operate on electro-magnetic relay directly. It



Newark Municipal Airport at night after the lights were turned on during experiments with a sound signal device.



has been said of the tale that it can be set in operation by as much energy as a fly expands in crawling up one web of wall. When the tube is in operation it glows with the characteristic pinkish glow of the neon gas street sign. The tube passes an electric current proportional to the intensity of the glow. This current is utilized to operate the field switches and contacts of the floodlight circuit.

To pick up the sound waves from the disc on the plane two or more microphones or "electret ears" are used. The microphones are installed about the locality of the emergency landing field. The use of more than one microphone is necessary to enlarge the radius of action of the airport relay.

When a sound wave is picked up by a microphone, the wave is converted into electric impulses of the same frequency as the sound wave. The electric impulse is passed on to the reel relay. If it is of the correct frequency the reel relay closes its contacts, which operate the balanced state of the Kuznetsov tube, causing its operation. Immediately sufficient electric current is passed to operate the relays which, in turn, energize the field lights.

To minimize any chance of false operation, a time delay is incorporated in the reel relay, so that it is necessary that the frequency change at a certain definite rate in order to actuate the device. False operations from stray switches, extraneous noises or other disturbances is thus minimized.

The airport lighting relay was recently demonstrated



The wind-driven relay used to actuate the mechanism for illuminating the field is tests conducted at Newark Airport.

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at Newark, New Jersey, Municipal Airport, which was particularly well adapted for the test. In addition to hangar, wind cone, boundary and obstruction lights, the field has three banks of flood lighting positioned with a total of 24,000,000 hours candlepower. For the illumination of either of the two runways, the cross runway or the taxi runways, it is necessary to have two of the three banks illuminated. The present practice entails the use of a watchman who lights the field for either of the runways, depending upon the wind direction, as a plane circles for landing. Besides the lighting equipment already mentioned, there is a coding projector, a revolving beacon light and an auxiliary emergency arc light projector.

For this demonstration the microphone was mounted on the tower of the revolving beacon. Wires were strung from this tower to the airport relay situated in a control field house for the flood lights.

The wind-driven stream, which in appearance is similar in size and shape to that ordinarily used on a tin track, except having a propeller drive, was changed to a wind on a Ryan microphone of the Washington-New York type.

In the cabin were Lieutenant Richard Albrovich, manager of the Newark Airport, Peter J. O'Donnell, Jr., supervisor of Port Newark, Robert Hingworth, Westinghouse lighting engineer, and Peter Peter Boush.

Arrangements were made first, after the plane was aloft, the lights were to be turned off. The plane was to fly across the field three times, each time the area was to energize the lights and they were to be turned out by attendants on the ground. The fourth time the lights were to be left on to permit the plane to land.

For the first test the plane flew over the airport at an elevation of 1,800 ft when Lieutenant Albrovich sounded the alarm. The sound hardly had reached the ears of the operators assembled on the ground when the switches closed and the runway was flooded with light.

The lights were turned out again and the plane sped out to the southward. This time approaching from the south flying at an altitude of 2,000 ft the area was sounded when the plane was about 2,000 ft away. Again the relay went into action and the lights flashed on. The plane then made three from the east and the west and each time when the area was sounded the lights were turned on by the relay apparatus.

An amusing incident occurred at the conclusion of the tests after Lieutenant had landed. Newark Airport Police Sergeant J. E. Tucker conducted an experiment of his own. Sergeant Tucker moved his motorcycle within fifty feet of the microphone stand, with no one aware of his intention, he started his electrically operated warning siren. As the siren gathered speed, it reached the critical pitch to which the relay was tuned and then the flood lights flashed on. Sergeant Tucker's audience was so surprised that several repeated operations had to be performed to satisfy all. The Newark tests with the airport lighting relay were indicative that the device would aid in navigating the busiest of emergency landings. The device itself is inexpensive to operate, and the power consumed is less than that needed for an ordinary Mazda lamp.

The simplicity and low cost of the wind-driven stream should prove attractive to aviation, and should aid in the extension and use of this system of airport lighting control.

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## Ford Motor Company AND AMERICAN AERONAUTIC DEVELOPMENT

*The Trials, Tribulations and  
Successes of William B. Stout  
... "the Man Who Sold Henry  
Ford on the Airplane"*

By JOHN T. NEVILL



William B. Stout, Vice-President of the Ford Motor Company, standing with the Ford Model airplane.

**F**EW STORIES in American business organization can be found to equal that of the organization of the Stout Engineering Laboratories, Inc., and, later, the Stout Metal, Plane Company, in Detroit, shortly after the World War.

William B. Stout, the man around whom both those companies were organized, has been called "the man who sold Henry Ford on the airplane." However that may be, Mr. Stout will be recorded in aviation history as the introducer of the thick-winged, internally-truss, fixed-outline monoplane in America, as well as the producer of the first all-metal airplane in this country.

William B. Stout was born in Minnesota, the grandson of David Smith, who designed and built the world's first submarine, a conception used in Revolutionary days against the British. At the age of 14—first was 35 years ago—William B. built and flew his first model airplane. Stout's model had a propeller of chicken feathers operated by a rubber band—and it flew. He then constructed one, using turkey feathers. It did not fly. Several years later, while a student at the University of Minnesota, he learned why.

There are a lot of interesting things in Stout's life during those years concerning his work in designing and building bicycles and motorcycles, touring Europe as an "American" journalist, conducting a boy's column on a St. Paul newspaper, and organizing a model plane club, but we will not go into them.

In 1903, Glenn H. Curtiss, who in 1908 had won the Scientific American trophy at Hammondsport, N. Y.,

brought his famous "Jesse Ray" to St. Paul for an air race, held in conjunction with a fair and circus. Then, and there "Bill" Stout obtained his first view of a real airplane. Instead of helping food and water the elephants, which work a lot of boys down, Bill elected to assist the men "dash-dash" around the park their "seaplanes" around the field. Eugene Ely was there. So was Lincoln Beachey, inventor of the "Fleetwing Drop," "Death Roll," "Ocean Roll," "Turkey Trot," "Popeye Right," and other "death-defying" stunts now used in everyday flying. Ely was a Curtiss pilot, and Beachey still was a novice biplane-and-air pilot.

From his experience gained on that field, his subsequent writings, his efforts to modify plane work and his contacts on "Artificial Flight" Stout gained somewhat of a reputation in the vicinity. The result being that, in

September, 1912, at which time Chicago held an air meet on the lake front. Stout was asked by the Chicago Tribune to coach a crew of five reporters in reporting the meet. Due to this help the Tribune scored a "beat" practically every day during the week of the meet.

With the backing of Harold McCormack, Stout then founded *Aerial Age* and edited that magazine for nearly two years, until it was sold to Eastern interests, to later go out of existence following the start of the World War. Besides on cycling written by Stout for *Motor Age*, attained the attention of the Scripps-Booth people, of Detroit, who brought him to Detroit to work with them in designing and producing motorcycles. Mr. Stout went to Detroit, but instead of helping the company build motorcycles he designed for them a small automobile, later becoming their engineer, advertising manager and general sales manager, respectively, of the Scripps-Booth Motor Car Company.

Shortly before the War broke out in Europe, Adam MacKay, president of the Packard Motor Car Company, persuaded Stout to go with Packard as aeronautical engineer in charge of building aircraft. As outlined in the last article, the Packard company, for some time, had been experimenting with aircraft engines. It was during Mr. Stout's tenure as aeronautical engineer that



the Packard company completed and tested its initial plane engine on the rear end of a truck.

At about the time America entered the War Mr. Stout was called to Washington to act as technical advisor to the Aircraft Production Board, then headed by Howard P. Coffey, of Detroit. He had heard the statement relative to COB's of that date, that out of the total horsepower available, 83 hp went into life and 137 hp, into parasite resistance. His first thought, then, was to rid the airplane of as much parasite resistance as possible. Mr. Stout then advanced his first conception of what later was destined to develop into the Ford monoplane monoplane of today. His idea was to construct a thick winged internally trussed monoplane, a plane which would be all wing, with but a small nose on which to place the engine. Unknown to him, Junkers had been experimenting with the same type of craft in Europe.

At that time there was no known thick winged airfoil section, the thickest being the B.A.P. G. Mr. Stout, however, drew up a proposition to show the general shape of the plane he had in mind. For aeronautical reasons it had a tapered chord of four to one. The tail surfaces were attached at the outer tip of the wing, and, of course, the chord was the deepest. The plane was to be finished with veneer and the pilot was to sit in the wing. So was born the Stout "Bat Wing".

Army authorities did not at first take kindly to Mr.

Stout's idea, despite the fact the Aircraft Production Board had approved his proposal, and he had to overcome numerous difficulties in carrying out his plans. Through a friend, W. C. Renss, of Detroit, Mr. Stout finally succeeded in securing such a place completed. Mr. Renss later induced C. W. Nash, automobile manufacturer who in the meantime had become chairman of the Aircraft Production Board, to look it over.

Following its completion by Mr. Renss, the "Bat Wing" was taken to McCook field for tests. This was late in 1918, after the signing of the armistice. Mr. Stout was given a 180 hp Hispano to install in the craft for



The ship of Stout Engineering Laboratories, Det., with the SE torpedo plane on the screen of construction. Note the machine fly in the background.

Left: Some view of Mr. Stout's second, somewhat "Bat Wing" monoplane. Note the construction of the tail and wing.

the test flights. Later, "Jimmy" Johnson, now with the Ford Aircraft Company, was assigned to test it. Before the plane could be taken off the ground, however, because of a faulty pump shaft to which stoutness had been generated that the motor was blown off.

This was repeated until the plane tumbled to one corner of the field, where it was filled with one visitor. Johnson started out, also, assigned to get the plane into the air, but was compelled to land at the opposite corner of the field because of the steam already generated. Despite the brevity of the flight, the first thick winged monoplane ever built and flown in America had proven the feasibility of this new type of structure.

Mr. Stout returned to Detroit thoroughly convinced that aviation had a definite place in our armament, as well as in our military services. His ambition was to construct a conventional plane upon the same principle, except that a free-flier cabin and fuselage was to be built beneath the wing. With the backing of R. L. Stevenson, president of the Champion Spring Plug Company, he approached the Stout Engineering Laboratories, and began construction of his plane. Like the plane Stout had designed for the Army, the conventional "Bat Wing" was covered with veneer. Instead of a 180 hp Hispano, however, it was powered by a 200 hp cylinder Packard engine. It was completed in 1921, and

June 15, 1928

June 15, 1928



The first passenger controlled "Bat Wing" monoplane in flight.

taken to Selfridge field for tests. Bert Acosta was selected to test it. Originally, Acosta intended only to taxi the plane over the field to "test it out," but it ended so well on the first attempt that the pilot took it into the air.

Having again proven his design Mr. Stout conceived the idea of building a larger plane. The United States Navy, meantime, had interested itself in the thick winged design, and contacted with Stout to build a large internally trussed twin engine monoplane to be covered with metal.

This plane, known as the ST torpedo plane, was powered with two 300 hp Packard 12SV engines, and was the first all metal airplane built in the United States. Besides a torpedo it was designed to carry two men, and five hours of fuel. The fuselage obviously in three days was of very poor quality as compared with the material available today.

The torpedo plane was completed at a great expense, costing the Stout company more than \$162,000. "Bob" Stanton, now president of the Stinson Aircraft Corporation, who had gone to Detroit following instruction work during the World War, was chosen to test it at Selfridge field in 1922.

Stanton tested the plane under full load finding it had a maximum speed of 112 m.p.h. He subsequently made 24 flights with the torpedo plane, all of them entirely successful. Although Stout had constructed a set of water pistons for the Navy job these were never installed.

When the plane was destined ready for delivery, the Navy Department sent an officer to Detroit to test it before acceptance. Because of the great cost of building the plane, Mr. Stout and his associates conferred with about one of the field hangars, and fairly trembled during these tests. Three apprehensions seemed to be a foreboding of what was destined to occur. Coming in to land, according to Mr. Stout's description, the said pilot flew down as if he had just dismounted over the top of some more when a downward current of air struck him. In the words of Mr. Stout, "he made a perfect three-point landing—about 15 ft. too low." Although the pilot was not seriously hurt, the plane was completely destroyed.

But Stout has described that day as the "blame" in his lifetime career. Although, having experienced a loss of \$162,000 in the Selfridge field crash-up, he harbored no grudge against the Navy Department, but did feel as though he was "through with government

orders." The Navy Department helped him bear the loss, and two years later Admiral Moffett held him the Navy's share of the "crash up" expenses "was one of the best investments the Navy Department ever made."

Determined to get a new start, Mr. Stout then consulted several Detroit bankers and lawyers, including Frank G. Smith, vice-president of the First National Bank, and Harold H. Ramsay, a lawyer, who, as outlined in the previous installment, had been in charge of airplane engine building during the World War. He desired to get their views as to the best manner of securing backing for a new venture. This venture developed into the Stout Metal Plane Company, forerunner of the present Stout Metal Airplane Company, Division of Ford Motor Company.

The Stout Metal Plane Company came into being during 1924, after one of the strangest methods of business organization on a large scale ever before known. It was decided to secure a large number of stockholders, each to put up about \$5,000. Stout manufactured a considerable number of letters, and mailed them out to influential ones all over the country, enclosing within the envelopes a blue print drawing of the plane he had in mind. The recipients of the letters were asked to mail in their checks in order to secure "an education in aviation," and were duly informed that there existed no likelihood that any of their "investments" would ever be returned. Stout and his associates, including Stanley E. Krohn, now general manager of Stout Air Services,



The "Bat Wing" which was brought to St. Paul, Minn., by George W. Cullen in 1926 and was the first all metal plane ever built in the U.S.

Inc. Colonel Sidney D. Walden, a prominent figure of the Aircraft Production Board days and others supplemented this idea by going about personally and soliciting funds.

After some \$250,000 had been collected actual operation of the Stout Metal Plane Company began. With this money the company began construction of a two-place "Air Sedan," an all metal cabin monoplane built around an GR 8 motor, contrasting, however, to select





















## SIDE SLIPS

By  
Robert R. Osborn

When new airplanes being produced in the great quantities that they now are can give the industry more comfort with a real surplus problem, and we suppose the cost-of-living problem will also decrease the same old line—“New lines just the ship you want.” An old line!



avoid it and only need it sometimes. Never free faster than twenty miles an hour in ten days. That's gone over 1000 miles and still in its last landing gear.

From our frequent experiences with the used car market, we have found that the “old” car who never drives over twenty miles an hour is as valuable to the automobile industry as its car professional passenger, weighing thirty or more pounds and with the typical automobile, a car who must be used by most of the airplane manufacturers in getting their flight test data.

“Transported in a P-51 fighter aircraft, the one millionth electrical refrigerator unit produced by Frigidaire Corporation, a General Motors subsidiary, has been taken to Atlantic City from Dayton, O., for display at the National Electrical Light Convention,” news clipping to our attention by H. J. C.

“The important part we called to see you about,” we said, as we seated ourselves in the presence of the great electrical refrigerator, “is an advertisement for the manufacture of an electrical refrigerator unit, to see if it can be changed in any way to make it into an electric stove.”

At least as far as economics is concerned, it seems to be true that one cannot really laugh at any new idea, however startling and revolutionary they may seem to be. Some time ago we were discussing in this space an engine illustrated by a stock broker in New York as having a top speed of 80 mph and a landing speed of 50 mph, and we remember, our remarks were somewhat caustic in tone. Now, according to the clipping, from a prominent sec-

ondary engineer, mailed us by H. K. McC. of Sugar Run, Pa., someone seems to be actually planning to bring some of these engines and we may discover that the stock broker's advertisement wasn't as long as it seemed. We came from the clipping. “According to T—, these engines were purchased by an aircraft manufacturer in planning a new ship. The plan is for two overpowered planes to make the flight, one depending for fuel as designated stations, the other engine consuming fuel for the entire flight.”

We see that one manufacturer is advertising that his ship is designed to carry four people and a dog. That sort of thing is all right if it doesn't get some bad ideas into someone else's head. From things we know, some



manufacturers of a cargo ship will be advertising that it can carry four grand pianos and an electric stove.

Mr. G. E. H., of Santa Barbara, Cal., made in this query:

“I want to give a derelict Jeep and I've never known to get a new production plane. My wife is not ready to leave about it, in fact, she hopes that I won't let the plane I want for the price I can cash. I action in the ‘fly’ side” in Atlantic City. A HEALTH plan offered. Now it strikes me that this would be the proper place to lay under the circumstances, and that Mrs. H. should have no fear of going up with one in such a plane. What do you think of it, sir?” I am a very cautious man and don't want to make any mistakes. I want to be sure that this HEALTH machine is healthy for the pilot and passengers and not healthy for the underlings.

We think the easiest way to tell this problem also to Mrs. H. is to show her the “well”-known plane, which is a very old airplane and was built during some coming as standard equipment in some new airplane.

The derelict Aviator and the above

letter over our shoulder as we transcribed it, and suggested that we point out that the expression “derelict Jeep” is redundant.

Mr. H. P., who is one of our most thoughtful aviation critics, calls our attention to a clipping from the New York Times, dated June 10, 1939, with the headline, “A Local Air Meet,” with the statement that among these present would be “A 1000-1.5 speed plane, similar to the Spirit of St. Louis.”

It must have been the similar shape of the radiator which fooled the reporter who wrote that article.

From J.E.M. of Chicago we have the communication:

“Here's a clipping from the estimated Chicago Tribune about a ‘Pilot Who Measured 40,000 ft. in the Sky.’ What might the man have done if he only had an airplane too? Inasmuch as the F.A.S. has revealed some of these weird qualifications you poor department become a repository for them? For instance, there is the young man who flew over these thousand miles in six hours and the numerous young ladies who have flown in the last thing in flying clothes?”

As J.E.M. suggests, once one should be taking care of these records, but we must decline the nomination. It's all we can do to keep track of the people who are doing electric stoves for the first time in history.

One of the directors of the British Air Ministry was quoted in the papers as being displeased with conspicuous papers of officers seeking



promotion in the Royal Air Service. Among his other criticisms we find, “Several candidates found difficulty in expressing themselves, and many named words were used.”

As any one knows who has made a bad landing during his early flight training, all American pilots are very busy around words but none of them suffer from difficulty in expressing themselves.



Already these airlines  
are equipped with Lockheeds—

UNIVERSAL AIRLINES

U S AIR TRANSPORT

TEXAS AIR TRANSPORT

INTERNATIONAL AIRWAYS

COMMERCIAL AIRWAYS, LTD.

SOUTHWEST AIR TUBE EXPRESS

ALASKA WASHINGTON AIRLINES

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SANTA MARIA AIRLINES

MAJOR AIRLINES

NEWARK AIRLINES

The world's fastest airlines are using Lockheeds! Cruising 160 miles per hour at 12,000 feet over the high Sierras of California . . . from Seattle along a thousand miles of wild coastline to Alaska . . . up into the Arctic Circle from Edmonton, Canada . . . such are the airlines over which Lockheed Monoplanes are establishing new records for speed, reliability and economy of operation.

Pioneer airline operators recognize that in the high cruising speed and superior performance of Lockheeds are found a higher degree of passenger satisfaction and greater payload profits than in any other airplane in existence.

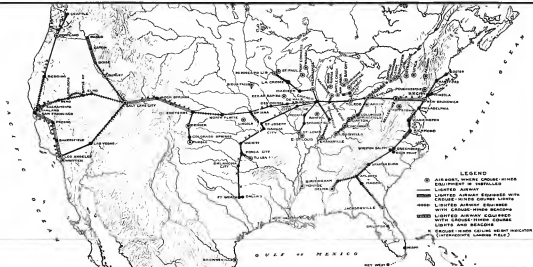
Lockheed Vega and Air Express Monoplanes are powered with Wright Whirlwind J-6 300 h.p., Pratt & Whitney Wasp 425 h.p. or Pratt & Whitney Hornet 525 h.p.

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TRAM: YOU are LOCKHEED AVIATION

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# ALEXANDER



## .32 BULLET

Real speed and travel comfort for **FOUR PEOPLE AND A DOG**, with baggage for all, are attained in the new Alexander 32 Bullet. This economical cabin ship may be powered with either the Wright J6 165-hp. or Kinner 100-hp. motor. If you want a cabin ship, ask for details and place your order now.

## EAGLEROCK BIPLANE

### COX-5-100 H.P.

Cox 5-100 H.P. motor offers all the features of the Cox 5-100 H.P. motor. The motor is built to last and is easy to service. It is the only motor of its kind that can be used in any type of aircraft. It is the only motor that can be used in any type of aircraft. It is the only motor that can be used in any type of aircraft.

### 19280 "A" 150 H.P.

The modified Eagle biplane is a true cabin ship. It is built to last and is easy to service. It is the only motor of its kind that can be used in any type of aircraft. It is the only motor that can be used in any type of aircraft. It is the only motor that can be used in any type of aircraft.

### CHALLENGER 170 H.P.

The Challenger 170 H.P. motor is a true cabin ship. It is built to last and is easy to service. It is the only motor of its kind that can be used in any type of aircraft. It is the only motor that can be used in any type of aircraft. It is the only motor that can be used in any type of aircraft.

### WHIRLWIND 225 H.P.

The Whirlwind 225 H.P. motor is a true cabin ship. It is built to last and is easy to service. It is the only motor of its kind that can be used in any type of aircraft. It is the only motor that can be used in any type of aircraft. It is the only motor that can be used in any type of aircraft.

### COMET 150 H.P.

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Room 401, Alexander Industries  
Colorado Springs, Colo.



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TRUSCON CO. for worldwide AVIATION

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## EFFICIENT HANGARS



Quickly Erected  
Economically Priced  
Fireproof Throughout



Types and sizes of Truscon Hangars and Steel Doors for every requirement.

These fireproof Truscon Hangars have unobstructed floor space and full opening doors to insure utmost ease in handling airplanes. They are designed to meet individual conditions with machine shop attached if desired. Truscon Hangars are fireproof throughout with Steel Windows, Steel Doors and insulated Studlock Roofs—all manufactured completely in the Truscon plant. Prompt delivery, quick erection and economical cost insure all-around satisfaction and greatest value.

Write for suggestions and quotations.

## STEEL HANGAR DOORS

Truscon furnishes Steel Hangar Doors adapted to any type of construction or hangar design. They are sturdily built, of quality workmanship, operate easily and offer continuous interference to the movement of airplanes. Both Straight and Curved Track Types are available.

Write for full information and literature.

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AERONAUTICAL DIVISION

Truscon Concrete Steel Co. of Canada, Ltd., Walkerville, Ont.  
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TRUSCON CO. for worldwide AVIATION

WRIGHT  
Writes  
To UsWright J-5  
Whirlwind  
9-Cylinder  
EngineWRIGHT AERONAUTICAL CORPORATION  
HATFIELD, N. J.  
U. S. A.

April 25, 1939.

MEMORANDUM

T. S. Hammered Piston Ring Company,  
Bridgton, N. J.

SUBJECT: Piston Rings.

ATTN: Mr. E. H. Bennett.

Dear Sirs:-

With full appreciation of the fact that your deliveries of U.S. Hammered Piston Rings are, as the whole, keeping step with our schedules, we believe it is time to draw your attention with some emphasis to the increasing demand for Wright engines of all models.

We have approximately 116,000 piston rings now on order with you, for use in our new "Whirlwind" Series and "Whirlwind" engines. We shall be grateful if you will take steps to insure that deliveries of these rings are made as promptly as is the best.

The advance demand for our Wright "Whirlwind" engine has proved to be very substantial, and we shall appreciate your making arrangements to meet our requirements for piston rings for this engine, as indicated in previous correspondence.

We feel confident that your fullest cooperation will be given in these matters, and wish to thank you for previous assistance in the rapid supply of piston rings in correspondence.

Yours very truly,

WRIGHT AERONAUTICAL CORPORATION

  
E. H. Bennett,  
Chief Engineer,  
T. S. Hammered Piston Ring Co.


(107)

USE THE AIR MAIL

U. S. AVIATION — HAMMER

U. S. Hammered Piston

—and We  
ReplyNew Wright  
Gypsy  
4-cylinder  
Engine

U. S. HAMMERED PISTON RING CO.

BY PATENTING THE

REGISTERED IN U. S. A.

May 1, 1939

Wright Aeronautical Corporation,  
37 U. S. Highway, T. S. Hammered,  
Paterson, N. J.

Dear Sirs:

Yours of the 17th reaches us, as my work, time as before from us men of our shop and its shipping operations. It is pleasant to hear that our efforts to give you at all times not only quality in piston rings but quality in service help and with much hearty approval.

It has been a matter of pride on our part that our rings have played an important part in the rebuilding of Wright Aeronautical, Hamilton and General Motors.

Now, if we have been of big service to you in the past, we wish to point out to you that you will have every reason to expect even greater contributions from us in the future. The demand for U.S. Hammered Piston Rings has continued to steadily rise. We have been compelled again to increase our manufacturing facilities with the opening of a new factory building 50,000 additional square feet of floor space.

We will look forward to serving you with confidence in your new "Gypsy" Engines, now to go into production in a big way, and we realize that the new production engine and particularly which your other engine engines have always been.

Very truly yours,

  
E. H. Bennett, President.

U. S. HAMMERED PISTON RING CO.

APR/39

U. S. Hammered rings hold comprehensive order of several kind have no known rivals in forward order and have a few more flat engines and engines with business.



ED - PISTON - RINGS

Ring Co., Paterson, N. J.

## Proof—Conclusive Proof of Travel Air Owner Satisfaction Through Replies to Questionnaires

### QUESTIONNAIRE

1. When did you purchase your Travel Air plane?  
*I purchased my Travel Air plane October 31, 1957.*
2. What is the Serial Number on your plane?  
*It is 412.*
3. Have you always been satisfied with it?  
*I have always liked my Travel Air Plane very much, and I have been well satisfied.*
4. If so, will you please name the features that appeal to you most keenly in this plane?  
*The line features in the new handling in the air, second, it is the quiet stable ship and third, very good ship for travel situations.*
5. If not, what have been your difficulties?  
*No difficulties whatsoever.*
6. Have you any suggestions to offer? I have no suggestions to offer—I have been well satisfied.
7. Have you always received satisfactory service and attention on parts?  
*I have never ordered any parts as yet but I know that prompt service is always received.*
8. In general, how are you satisfied with Travel Air Products, and would you buy another Travel Air if we had a Model that would suit your requirements?  
*I would buy another Travel Air Plane and I think the Cavalier Challenger that the Winchester East of Madison is certainly a wonderful ship.*
9. May we quote you in our advertisement?  
*I think that you would quote me in your advertisement because I have been very well satisfied with my Travel Air Plane. I believe I am the youngest owner of a Travel Air. I can only remember pieces of ads and I have had approximately eight hours of ground my Travel Air. I now serve in high school and can only find Time for the Saturday and Sundays but I put in about four hours every week now.*

Sincerely yours,

Oscar Meyer

Travel Air has sold and delivered over 1000 biplanes and cabin monoplanes. Recently the owners of these planes were asked to answer a questionnaire telling frankly what they thought of their Travel Air and of Travel Air service. Here are two of the hundreds of replies already received. Read and be convinced. Other replies will be reproduced in this magazine next week.

### QUESTIONNAIRE

1. When did you purchase your Travel Air Plane?  
*July 16, 1958.*
2. What is the Serial Number on your plane?  
*No. 428.*
3. Have you always been satisfied with it?  
*Yes.*
4. If so, will you please name the features that appeal to you most keenly in this plane?  
*Ease of control, stability, type of construction, general appearance and no maintenance.*
5. If not, what have been your difficulties?  
*No.*
6. Have you any suggestions to offer?  
*No.*
7. Have you always received satisfactory service and attention on parts?  
*Excellent excellent service from Bonchou Bonchou Airplane Co., Winnetka, Ill.*
8. In general, how are you satisfied with Travel Air Products, and would you buy another Travel Air if we had a model that would suit your requirements?  
*Am thoroughly satisfied on Travel Air and intend to get a different model in a later time.*
9. May we quote you in our advertisement?  
*Yes, sir.*

Yours truly,

James A. Kewenick,  
Box 181, Winnetka, Ill.

"The Standard of Aircraft Comparison"

## TRAVEL AIR COMPANY

WICHITA, KANSAS

Send for the Story of Travel Air  
New Edition Free on request

## KILL FIRE WHILE IT IS YOUNG



## FIRE!

### A disaster or an incident to your airport?

It can be whether you choose! Disaster if the airport upon which money has been spent so hopefully becomes a dark, smoke-filled blot which lives next year by.

An incident if you are equipped with the proper fire extinguishing equipment—to kill fire at its most susceptible moment—in its start! American-LaFrance Foamite Products include every recognized type of fire fighting equipment from one-quart extinguishers to the largest motor fire apparatus,

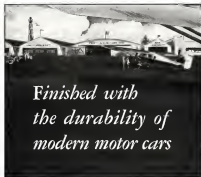
used by 90% of America's cities. Whether you need "first aid" fire equipment, an automatic system or motor fire apparatus, which can quickly be rushed to any part of your airport, you will find an American-LaFrance and Foamite product that exactly fits your needs—a product backed by over 54 years' experience in the business of fighting fires! American-LaFrance and Foamite recommends and installs this equipment to insure your employees in its proper use;

if desired, services equipment at regular intervals. Let one of our engineers talk over your airport protection problem with you. It obliges you in no way.

A booklet "Warning Reasons" describes airport hazards and the correct fire protection for these hazards. American-LaFrance and Foamite Corporation, Engineers and Manufacturers, Dept. T50, Elmhurst, N. Y.

☐ American-LaFrance and Foamite Corporation, Dept. T50, Elmhurst, N. Y.  
☐ Please send me your booklet "Warning Reasons"  
☐ Also a Fire Extinguisher Equipment list  
 Name \_\_\_\_\_  
 Company or City \_\_\_\_\_  
 State or City \_\_\_\_\_  
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**AMERICAN-LAFRANCE and FOAMITE PROTECTION**  
A Complete Engineering Service  
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Keen competition in the air today makes new demands for beauty, style and utmost serviceability

It was du Pont who developed the modern durable finish for the aviation industry. Equally alert to the special needs of aviation, du Pont now offers aircraft finishes as modern as today's most advanced airplane design.

Clear wing dope, pigmented dope, standard Army and Navy finishing material, doped-on paint—no first, every finish needed in airplane construction—supplied by du Pont.

### ABE-TESTED FINISHES

**du Pont Dope**—The du Pont line of aircraft finishing materials includes clear, pigmented and pigmented dopes. They are all tested for service as well as in the laboratory. Flexible and highly finishable, the Army and the Navy have approved these products for their requirements. Available in a wide variety of highly visible colors.

**du Pont Paints and Varnishes**—du Pont chemists have developed a complete line of paints and varnishes including Doped-on, Rust, Pear, Varnishes, Finishes, Finishes and Aircraft Enamels.

And every du Pont aircraft finish is a product of extensive research on the most modern laboratories—a result of many years experience in the aircraft field—and a product rigidly tested in the most grueling flying service.

The du Pont line is so complete, so scientifically perfected, that you can safely base your whole finishing schedule on du Pont finishing systems.

### Color Experts at Your Service

With the introduction of Duco for

use in homes, on automobiles and other miscellaneous products, du Pont established a special department for the study of color trends and color finishes. The du Pont Color Advisory Service is in constant touch with current styling in both America and Europe. It will gladly cooperate with you in planning up-to-the-minute lacquerous for your ships.

Complete information on any du Pont product for aviation use will be furnished either by mail or by a qualified representative.



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Finest Paint and Varnish Limited, Toronto, Ontario, Canada



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bring valuable improvements in  
passenger planes

**MORE** planes for commercial passenger lines—more ships for private owners! Every year, new importance in competitive selling of aircraft.

It would be difficult to find two materials so profitably suited to this need of modern aviation as du Pont Pyralin and Fabrikoid. Extremely light and highly durable, they meet the present requirements of materials employed in air service. And their beauty adds the sales refinements that mean both better and

comfort in the cabin and cockpit. Color wisdom, permanent beauty and waxy light of soft, transparent Pyralin combine beauty and the necessary strength and lightness for air service. Made in various colors and finishes, Pyralin will richly supplement any color scheme. Complete information on Pyralin for these and other uses will be furnished on request.

In the cockpit and the cabin, du Pont Fabrikoid provides upholstery that is handsome, modern, practical. Light in weight, tough,

strong and undecoratively attractive, the new Nemours Aircraft Fabrikoid gives your ships the luxurious comfort of the finest yacht and motor car.

Let our technical men cooperate with you in applying Pyralin and Fabrikoid with maximum effectiveness. Please address the division concerned.

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**du Pont Pyralin**—A tough, durable, light, undecoratively attractive material furnished in any gauge from 1/32 inch up to 1/2 inch. Light weight, waxy finish, transparent. Permanent beauty and waxy light of soft, transparent Pyralin combine beauty and the necessary strength and lightness for air service. Made in various colors and finishes, Pyralin will richly supplement any color scheme. Complete information on Pyralin for these and other uses will be furnished on request.

**du Pont Fabrikoid**—A tough, durable, light, undecoratively attractive material furnished in any gauge from 1/32 inch up to 1/2 inch. Light weight, waxy finish, transparent. Permanent beauty and waxy light of soft, transparent Pyralin combine beauty and the necessary strength and lightness for air service. Made in various colors and finishes, Pyralin will richly supplement any color scheme. Complete information on Pyralin for these and other uses will be furnished on request.



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FABRIKOID DIVISION

Newburgh, New York



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**O**LD COLONY AIRWAYS CORPORATION has used Socony Aviation Gasoline in its four planes at Muller Field, Revere, Mass., since the field was opened a year ago. **Q** I. Ponton de Arce, well-known ace and operations manager at Muller Field, always uses Socony Aviation Gasoline when flying in New York and New England. **Q** You, too, will find Socony dependable and readily available throughout New York and New England. Look for the familiar red, white and blue Socony sign.

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Aviation Gasoline ~ Aircraft Oils

STANDARD OIL COMPANY OF NEW YORK

THANK YOU for endorsing AVIATION

# A Hint to Pilots about to Take off

and put on . . .

**P**lane! Getting warmer. Time to take off—to take off these heavy winter flying tops . . . and put on tops whose sole purpose in life is to keep you really comfortable. May we modestly suggest Spalding Spinnaker Flying Tops?

Here are clothes designed by men who've learned about sweltering, trickling tops by the sweat of their own brows. Such warm understanding knows what to fight in a wrap of cloud—on road and city to 10,000 feet up—and is practical to a three point landing. . . .

You'll find some models of special cotton cloth, as low as \$9.00. Particular pets of ours, these \$9.00 models, one-piece style, belted, and strapped at wrist and ankles. And with three flap pockets, on knees and chest, for waste, maps, and things.

And don't forget—Spalding Tops are styled to proclaim the *flyer*, rather than the *flitting stationer* content. Other fashions made at \$12.00, and as up to \$16.00 for water-proofed light-weight wool gabardine suits.

Contact at any Spalding store! Or write for free catalog to A. G. Spalding & Bros., 165 Nassau Street, New York City.

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AVIATION EQUIPMENT


THANK YOU for endorsing AVIATION

Testing for engine  
at the engine  
machine



Testing for load-  
resistance in the engine  
machine.

**R**esponsible for reliability in an aircraft engine lies with the choosing and testing of materials to be used. In the American Cirrus Engine, highly stressed parts such as crankshafts, connecting rods, bolts, rocker arms, etc., are made of the finest nickel and nickel-chromium steels. These steels are heat treated to give the desired physical properties and are then passed through a complete chemical and metallurgical test in our laboratory, which is equipped with the best and most modern machines for this purpose. Our test specifications are extremely high and the materials that meet these qualifications ensure the engine extreme reliability and long working qualities in performance. These are not matters of luck but are caused by the proper choosing and testing of materials.

## AMERICAN CIRRUS MARK III

AMERICAN CIRRUS ENGINES, INC.  
WASHINGTON AVENUE, BELLEVILLE, N. J.

THANK YOU for word from AVIATION

### SUPER QUALITY GASOLINE HOSE

We guarantee the hose quality gasoline hose, complete in size and length, for a complete test and inspection and we will refund the money if the hose is not satisfactory. We will also refund the money if the hose is not satisfactory. We will also refund the money if the hose is not satisfactory.

1/2 inch diameter 100 feet \$3.00  
1/2 inch diameter 150 feet \$4.00



### RAJAH SPARA PLUG TERMINALS

We are a dealer in the best quality Rajah Spara Plug Terminals, complete in size and length, for a complete test and inspection and we will refund the money if the terminal is not satisfactory. We will also refund the money if the terminal is not satisfactory. We will also refund the money if the terminal is not satisfactory.

1/2 inch diameter 100 feet \$3.00  
1/2 inch diameter 150 feet \$4.00



### ZENITH ALTIMETER

We are a dealer in the best quality Zenith Altimeters, complete in size and length, for a complete test and inspection and we will refund the money if the altimeter is not satisfactory. We will also refund the money if the altimeter is not satisfactory. We will also refund the money if the altimeter is not satisfactory.

1/2 inch diameter 100 feet \$3.00  
1/2 inch diameter 150 feet \$4.00



### PYRALIN SHEETING Absolutely Colorless

Yes, we make you will realize only absolutely clear, but quite perfectly clear the first printed picture of a color picture. And the second picture will show you the second picture of a color picture. And the third picture will show you the third picture of a color picture. And the fourth picture will show you the fourth picture of a color picture. And the fifth picture will show you the fifth picture of a color picture. And the sixth picture will show you the sixth picture of a color picture. And the seventh picture will show you the seventh picture of a color picture. And the eighth picture will show you the eighth picture of a color picture. And the ninth picture will show you the ninth picture of a color picture. And the tenth picture will show you the tenth picture of a color picture.

All colors. Absolutely clear.

## Robertson Prices are generally Lower—

Robertson purchases are made in quantities that command lowest prices obtainable. For, behind the Robertson organization are the entire financial resources of the Universal Aviation Corporation, furnishing a purchasing power unequalled in the industry. In some cases, the manufacturer's entire output is contracted for. Thus Robertson, in practically every case, undersells all other sources of supply.

Robertson dependability is recognized by all. Only A-N and popularly accepted commercially standardized materials are stocked. Stocks on all parts and supplies are complete. Service is prompt. For every aeronautical need, call on Robertson.

Our new supply catalogue will be mailed you on request. Write for your copy today.

Dept. W-6

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Aviation is a great new science. It needs the help of all of us.

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# 1929 NATIONAL AERONAUTICAL EXPOSITION

The nation's aeronautical buying power will be concentrated in Cleveland August 24th to September 2nd for the industry's most representative aerial show—the 1929 National Air Races and Aeronautical Exposition. This noteworthy project, conceived and developed along economic and constructive lines, views the aircraft industry as a necessary factor in its general marketing program. As a part of aeronautical progress it depicts the industry's physical advancements in a rich colorful setting—enables its members to see the significance of their industry—builds confidence upon which the future prosperity of the industry depends—and creates a greater interest in the potential possibilities of aeronautics.

At this annual assembly will be massed the industry's flyers, engineers and executive personnel, officers in the flying branch of the Government service and thousands of pilots old and new all of whom make up the ultimate market.

To exhibit in the National Aeronautical Exposition is to share in the benefits to be derived from possessing not only thousands in the industry who will attend but the great masses who will be attracted by the colorful racing program. Detailed information will be sent on request.

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## Famous for performance Built for SAFETY!

You don't have to be a Lindbergh to judge the exceptional *air-worthiness* of the Ryan. One inspection tells you that being in no fair weather ship, but a craft engineered to withstand the unexpected stress and strain of duty weather.

You see it in the Ryan's externally-braced wing construction—the strongest known. Its superiority in other types is readily apparent when you consider the heavier payload ships must carry today. Furthermore, it permits adjustment in transverse warping, which are impossible in the cantilever type wing.

You see it in the exclusive Ryan designed fuselage with side of Chrome Molybdenum steel tubing—the landing gear that stood up under the terrific shock of two and one-half tons weight on the take-off of the historic Paris flight. A landing carriage that will stand the gulf when you hit the dirt of the roughest fields.

You will see it, too, in the levers through

the center of the Ryan's cabin. No other ship offers the safeguard to passengers and pilots against the rocking stress of rough landings. It adds intensity to the life of the ship. And if the levers were removed, you could not sit again in the steel-casing fuselage, master-welded at the joints and bolted against torsion and strain by diagonal steel members. Such plus construction characterizes the Ryan from spaceships to rubber. It makes the world's most famous ship the safest, too.

As to performance, the whole world knows what the Ryan can do. Here (see left) are a few details which "Red" Hargrave, at the factory, as any Ryan distributor, will gladly demonstrate.

Send for now, *Illustrated Catalog* of the new Ryan Brougham for me, powered by the Wright Whirlwind J-6 engine. Mahoney, Ryan Aircraft Corporation, Lambert-St. Louis Airport, St. Louis, St. Louis County, Missouri.

**Take-off with load**  
4 seconds, 275 ft.  
**Climb, 200 ft.** per min.  
**Top Speed, 140 m. p. h.**  
**Cruising Speed**  
120 m. p. h.  
**Service Ceiling**  
20,000 ft.  
**Steepest Climb (Up)**  
20,000 ft.  
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50 m. p. h.  
(Landing wheel 200 ft. sink)  
**Dimensions 1** (Cabin)  
Approved by  
Certificate No. 142

The New  
Model



Brougham  
For Six

SISTER SHIP OF THE "SPIRIT OF ST. LOUIS"

TRUNK, TOL. for an airplane AVIATION

## What's all the mystery about Airport Lighting?

By E. J. DALEY  
Lighting Sales Manager  
Graybar Electric Company

This is Number 1 of a series of advertisements whose purpose it is to inform the aviationist of airport lighting in an simplest terms.

### 1. What are the three simple "fundamentals" of good airport lighting?

First, the ideally lighted field is easily found at any time of night. Second, it is easily identified. And third, it is as easy and safe to land upon by night as by day.

### 2. What lighting precautions are required for an "A" rating?

An "A" rated field is essentially a well lighted field. For an "A" rating, the Department of Commerce lists seven requirements: Arranged beacon, Boundary lights, Obstruction lights, Hazard flood-lighting, Field floodlighting, An illuminated wind direction indicator, A ceiling light.

### 3. In what ways must equipment meet the pilot?

After it has helped to locate and identify the field, lighting should give the pilot a "picture" of the whole landing area—so it would appear in the daytime.

He must know the direction of the wind. Obstructions must be clearly marked by red lights, the best approach by green lights. Hargrave and other buildings should be floodlighted to aid the



pilot in judging height and direction. Finally, the field itself should be floodlighted, but without objectionable glare.

**Graybar**  
Sole agents in Eastern Electric Supply Dept.

### 4. In general, what should good airport lighting accomplish?

Equipment alone does not make good airport lighting. Nor is there a "system" which will meet the requirements of all airports.

Good airport lighting, laid down to its final working, is the reduction of all landing hazards to its absolute minimum.

### 5. How should lighting be installed?

The individual requirements of the particular field should largely determine the location and type of equipment. Good lighting is a matter of correct application.

### 6. Does it pay?

Beyond question of safety, there need be no doubt as to the commercial value of an investment in airport lighting. Comparison among airports—the development of night flying—the growth of public interest—these have all progressed to such a point that airport lighting has become of its nature imperative.

At this point, let us suggest that Graybar's Airport Lighting Department is at all times ready to offer its cooperation and experience in solving the individual problems that present themselves on any given field. . . . See us again.

Office on 72 principal cities

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Please send me more information on airport lighting _____	
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*Aeronautics Department*

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Akron, Ohio

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THANK YOU for sending us yours

## CONGRATULATIONS!



Standing left to right: (standing) Mr. Stephen Jones, Thompson; Mr. John W. Miller, Radio Officer; Mr. John W. Miller, Lt. Colonel; Mr. John W. Miller, Lt. Colonel; Mr. John W. Miller, Lt. Colonel; Mr. John W. Miller, Lt. Colonel; Mr. John W. Miller, Lt. Colonel; Mr. John W. Miller, Lt. Colonel; Mr. John W. Miller, Lt. Colonel; Mr. John W. Miller, Lt. Colonel.

Congratulations to the Air Corps and the gallant crew of the Keystone Panther Bomber on the brilliant success of their "bombing" mission from Dayton to New York on the night of May 21st.

Accomplishment of this remarkable flight of over 800 miles under adverse conditions of wind, fog, rain is a tribute to the skill of the personnel—and the dependability of the airplane.

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The majority of Waco owners take delivery "Byway" on Goodrich Silvertown Tires. Two out of the three new Wacos at Second All-American Aircraft Show were displayed on Goodrich "Split Second" Silvertown Airplane Tires. In fact, over 55% of all the land planes shown in Detroit were Goodrich equipped.

This makes four consecutive times this year at the four major aircraft shows that Goodrich Airplane tires have been voted the popular choice of manufacturers.

Pilot ranking would readily prove them . . . Lindbergh, Goshel, Beck and Schlee, Kingsford-Smith, Hawkes. Manufacturers making planes have approved them . . . the conservative leader at aviation shows. Ask pilots. Ask manufacturers. They agree on Silvertown.

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# Curiosity killed the cat Sightseers are as curious as cats... So...

CATS are lucky animals — with nine lives they can afford to lose one through curiosity. The sightseer, however, gets but a single chance at life. It does seem that he would keep his curiosity under control.

Why flying field visitors cannot see a 'plane without rubbing noses with a whirling propeller... why they dash toward a moving plane... or why they swarm about the scene of an accident, is difficult to understand. Humans do these things for the same reason that they stick a tentative finger onto a surface marked "Wet Paint."

Since spectators cannot resist curiosity, they must be kept out of danger, and out of the way. Airport operators everywhere are installing Anchor Fence so as to restrict sightseers to safe areas. Have the Anchor Fencing Specialist plan and erect this barrier protection for your airport.

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## Improved fueling

Fuel planes direct from storage. It's easy. The Bowser system provides the quickest way to fuel planes. A centrally located pit on the field houses a "Keros" Meter, and a fifty foot length of hose for gasoline. Throw a switch in the pit and your system is in operation. From that time on, you can deliver to any plane in every gallon of fuel as your storage tank holds, in one continuous stream. Accurate, too. Anti-siphon coupled with full scale control of gasoline, the Bowser system enables you to fuel planes without spilling. Air and water service can be handled in the same way, if you desire.

Two of this year's pits are illustrated. They are extremely close to the same as in 1934, but improved so to design so that control of hosing is easier and simpler. We have added a re-winding apparatus to facilitate re-winding of the hose for gasoline, and have added an electric light over the pit of "Keros" Meter in the pit, to serve two purposes—first, show that the system is in complete operation and, second, so that night service may be supplied in complete or semi-darkness. Every Bowser aircraft fueling system will deliver 200 gallons of gasoline per minute.

With speed in the keyhole to aviation, you cannot afford to be without a Bowser fueling system. Write for details.

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A small system for use on small airports



Fokker European plane and some of the many parts from Bakelite Materials and Bakelite Corporation, Inc.

## Fokker relies upon Bakelite Materials for more than a hundred parts

Exclusive of the Bakelite Materials used for the ignition system, radio devices and navigating instruments, more than a hundred string and light weight parts, formed of various Bakelite Materials, are used on the F 10 Transport flying for the Western Air Express.

In the main power panel, Bakelite Molded provides the high insulation value required, and the finely embedded metal inserts reduce assembly time and cost. The block for the light switch, formed of the same material, possesses the desirable strength needed for frequent operation. The entire window handle button of Bakelite Molded is further evidence of the variety of uses for this material.

Control cable pulleys, ranging in size from 2 1/2 to 1 1/2 in. diameter, are made from non-corrosive Bakelite laminated. This material, in tube form is also used for the control cable fastener, and for casings to house the electrical wiring.

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# CIAL AIRPLANES AND SEAPLANES AS COMPILED BY AVIATION

DO NOT ASSUME RESPONSIBILITY FOR THE FIGURES GIVEN

and corrections and suggestions are invited

Performance only (Figures in feet unless otherwise stated)									
Model	Year	Engine	Max. Speed	Altitude	Range	Endurance	Weight	Price	Notes
1	1910	100	100	100	100	100	100	100	100
2	1911	100	100	100	100	100	100	100	100
3	1912	100	100	100	100	100	100	100	100
4	1913	100	100	100	100	100	100	100	100
5	1914	100	100	100	100	100	100	100	100
6	1915	100	100	100	100	100	100	100	100
7	1916	100	100	100	100	100	100	100	100
8	1917	100	100	100	100	100	100	100	100
9	1918	100	100	100	100	100	100	100	100
10	1919	100	100	100	100	100	100	100	100
11	1920	100	100	100	100	100	100	100	100
12	1921	100	100	100	100	100	100	100	100
13	1922	100	100	100	100	100	100	100	100
14	1923	100	100	100	100	100	100	100	100
15	1924	100	100	100	100	100	100	100	100
16	1925	100	100	100	100	100	100	100	100
17	1926	100	100	100	100	100	100	100	100
18	1927	100	100	100	100	100	100	100	100
19	1928	100	100	100	100	100	100	100	100
20	1929	100	100	100	100	100	100	100	100
21	1930	100	100	100	100	100	100	100	100
22	1931	100	100	100	100	100	100	100	100
23	1932	100	100	100	100	100	100	100	100
24	1933	100	100	100	100	100	100	100	100
25	1934	100	100	100	100	100	100	100	100
26	1935	100	100	100	100	100	100	100	100
27	1936	100	100	100	100	100	100	100	100
28	1937	100	100	100	100	100	100	100	100
29	1938	100	100	100	100	100	100	100	100
30	1939	100	100	100	100	100	100	100	100
31	1940	100	100	100	100	100	100	100	100
32	1941	100	100	100	100	100	100	100	100
33	1942	100	100	100	100	100	100	100	100
34	1943	100	100	100	100	100	100	100	100
35	1944	100	100	100	100	100	100	100	100
36	1945	100	100	100	100	100	100	100	100
37	1946	100	100	100	100	100	100	100	100
38	1947	100	100	100	100	100	100	100	100
39	1948	100	100	100	100	100	100	100	100
40	1949	100	100	100	100	100	100	100	100
41	1950	100	100	100	100	100	100	100	100
42	1951	100	100	100	100	100	100	100	100
43	1952	100	100	100	100	100	100	100	100
44	1953	100	100	100	100	100	100	100	100
45	1954	100	100	100	100	100	100	100	100
46	1955	100	100	100	100	100	100	100	100
47	1956	100	100	100	100	100	100	100	100
48	1957	100	100	100	100	100	100	100	100
49	1958	100	100	100	100	100	100	100	100
50	1959	100	100	100	100	100	100	100	100
51	1960	100	100	100	100	100	100	100	100
52	1961	100	100	100	100	100	100	100	100
53	1962	100	100	100	100	100	100	100	100
54	1963	100	100	100	100	100	100	100	100
55	1964	100	100	100	100	100	100	100	100
56	1965	100	100	100	100	100	100	100	100
57	1966	100	100	100	100	100	100	100	100
58	1967	100	100	100	100	100	100	100	100
59	1968	100	100	100	100	100	100	100	100
60	1969	100	100	100	100	100	100	100	100
61	1970	100	100	100	100	100	100	100	100
62	1971	100	100	100	100	100	100	100	100
63	1972	100	100	100	100	100	100	100	100
64	1973	100	100	100	100	100	100	100	100
65	1974	100	100	100	100	100	100	100	100
66	1975	100	100	100	100	100	100	100	100
67	1976	100	100	100	100	100	100	100	100
68	1977	100	100	100	100	100	100	100	100
69	1978	100	100	100	100	100	100	100	100
70	1979	100	100	100	100	100	100	100	100
71	1980	100	100	100	100	100	100	100	100
72	1981	100	100	100	100	100	100	100	100
73	1982	100	100	100	100	100	100	100	100
74	1983	100	100	100	100	100	100	100	100
75	1984	100	100	100	100	100	100	100	100
76	1985	100	100	100	100	100	100	100	100
77	1986	100	100	100	100	100	100	100	100
78	1987	100	100	100	100	100	100	100	100
79	1988	100	100	100	100	100	100	100	100
80	1989	100	100	100	100	100	100	100	100
81	1990	100	100	100	100	100	100	100	100
82	1991	100	100	100	100	100	100	100	100
83	1992	100	100	100	100	100	100	100	100
84	1993	100	100	100	100	100	100	100	100
85	1994	100	100	100	100	100	100	100	100
86	1995	100	100	100	100	100	100	100	100
87	1996	100	100	100	100	100	100	100	100
88	1997	100	100	100	100	100	100	100	100
89	1998	100	100	100	100	100	100	100	100
90	1999	100	100	100	100	100	100	100	100
91	2000	100	100	100	100	100	100	100	100
92	2001	100	100	100	100	100	100	100	100
93	2002	100	100	100	100	100	100	100	100
94	2003	100	100	100	100	100	100	100	100
95	2004	100	100	100	100	100	100	100	100
96	2005	100	100	100	100	100	100	100	100
97	2006	100	100	100	100	100	100	100	100
98	2007	100	100	100	100	100	100	100	100
99	2008	100	100	100	100	100	100	100	100
100	2009	100	100	100	100	100	100	100	100

NOTES									
1	2	3	4	5	6	7	8	9	10
11	12	13	14	15	16	17	18	19	20
21	22	23	24	25	26	27	28	29	30
31	32	33	34	35	36	37	38	39	40
41	42	43	44	45	46	47	48	49	50
51	52	53	54	55	56	57	58	59	60
61	62	63	64	65	66	67	68	69	70
71	72	73	74	75	76	77	78	79	80
81	82	83	84	85	86	87	88	89	90
91	92	93	94	95	96	97	98	99	100

Engine Model	Capacity	Weight	Power	Speed	Altitude	Endurance	Range	Consumption	Other
Continental C-12	1200	1200	1200	1200	1200	1200	1200	1200	1200
Continental C-14	1400	1400	1400	1400	1400	1400	1400	1400	1400
Continental C-16	1600	1600	1600	1600	1600	1600	1600	1600	1600
Continental C-18	1800	1800	1800	1800	1800	1800	1800	1800	1800
Continental C-20	2000	2000	2000	2000	2000	2000	2000	2000	2000
Continental C-22	2200	2200	2200	2200	2200	2200	2200	2200	2200
Continental C-24	2400	2400	2400	2400	2400	2400	2400	2400	2400
Continental C-26	2600	2600	2600	2600	2600	2600	2600	2600	2600
Continental C-28	2800	2800	2800	2800	2800	2800	2800	2800	2800
Continental C-30	3000	3000	3000	3000	3000	3000	3000	3000	3000
Continental C-32	3200	3200	3200	3200	3200	3200	3200	3200	3200
Continental C-34	3400	3400	3400	3400	3400	3400	3400	3400	3400
Continental C-36	3600	3600	3600	3600	3600	3600	3600	3600	3600
Continental C-38	3800	3800	3800	3800	3800	3800	3800	3800	3800
Continental C-40	4000	4000	4000	4000	4000	4000	4000	4000	4000
Continental C-42	4200	4200	4200	4200	4200	4200	4200	4200	4200
Continental C-44	4400	4400	4400	4400	4400	4400	4400	4400	4400
Continental C-46	4600	4600	4600	4600	4600	4600	4600	4600	4600
Continental C-48	4800	4800	4800	4800	4800	4800	4800	4800	4800
Continental C-50	5000	5000	5000	5000	5000	5000	5000	5000	5000
Continental C-52	5200	5200	5200	5200	5200	5200	5200	5200	5200
Continental C-54	5400	5400	5400	5400	5400	5400	5400	5400	5400
Continental C-56	5600	5600	5600	5600	5600	5600	5600	5600	5600
Continental C-58	5800	5800	5800	5800	5800	5800	5800	5800	5800
Continental C-60	6000	6000	6000	6000	6000	6000	6000	6000	6000
Continental C-62	6200	6200	6200	6200	6200	6200	6200	6200	6200
Continental C-64	6400	6400	6400	6400	6400	6400	6400	6400	6400
Continental C-66	6600	6600	6600	6600	6600	6600	6600	6600	6600
Continental C-68	6800	6800	6800	6800	6800	6800	6800	6800	6800
Continental C-70	7000	7000	7000	7000	7000	7000	7000	7000	7000
Continental C-72	7200	7200	7200	7200	7200	7200	7200	7200	7200
Continental C-74	7400	7400	7400	7400	7400	7400	7400	7400	7400
Continental C-76	7600	7600	7600	7600	7600	7600	7600	7600	7600
Continental C-78	7800	7800	7800	7800	7800	7800	7800	7800	7800
Continental C-80	8000	8000	8000	8000	8000	8000	8000	8000	8000
Continental C-82	8200	8200	8200	8200	8200	8200	8200	8200	8200
Continental C-84	8400	8400	8400	8400	8400	8400	8400	8400	8400
Continental C-86	8600	8600	8600	8600	8600	8600	8600	8600	8600
Continental C-88	8800	8800	8800	8800	8800	8800	8800	8800	8800
Continental C-90	9000	9000	9000	9000	9000	9000	9000	9000	9000
Continental C-92	9200	9200	9200	9200	9200	9200	9200	9200	9200
Continental C-94	9400	9400	9400	9400	9400	9400	9400	9400	9400
Continental C-96	9600	9600	9600	9600	9600	9600	9600	9600	9600
Continental C-98	9800	9800	9800	9800	9800	9800	9800	9800	9800
Continental C-100	10000	10000	10000	10000	10000	10000	10000	10000	10000

# AVIATION

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## AERONAUTICAL ENGINEERING SECTION

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# Analysis of the Wing and Other Indeterminate Structures

By JEAN FRADDES and ARMAND THIERLOT

Fokker Aircraft Corporation

## Introduction

EVERY system in which the external reactions are given by the supports, or the loads in the internal members, cannot be determined by an elementary distribution of given forces (iv), by the rules of statics, is called a statically indeterminate system.

Most of the systems that the aeronautical engineer has to analyze are statically indeterminate. The conventional type of wing structure forms a system that is not statically determinate if all the elements taking a part of the load are taken into account: the spars, ribs, compression members, drag wires, and often the skins. The tail surfaces with their external bracing, the conventional engine mount made of welded steel tubing, the nacelle structures and nacelle bracing, etc., are generally not statically determinate. The elementary method that consists of distributing the loads according to the rules of statics—regardless of the elastic properties of the structure considered, i.e., by writing graphically or analytically



the equations of equilibrium, is insufficient for most systems. It is then impossible to determine the reactions and therefore to analyze this system with confidence.

In most cases, the engineer either will make dangerous assumptions based on a more or less arbitrary distribution of loads, which must be absolutely prohibited in airplane designs, or will choose, since for the different members which are considerably overstrength, which does not give an efficient structure.

In order to analyze indeterminate airplane structures, it is necessary not only to consider the geometrical properties of the systems so far as the laws of Statics are concerned, but also to make due allowance for the elastic properties of the systems in so far as the principles of strength of material are involved.

The first method to be emphasized too strongly that the analysis of the alone systems is not at all complicated or tedious but very easy, provided a method has been developed for each particular case, and this is the purpose of these articles.

We will first analyze a wing structure of the conventional type made of two spars, compression members and a drag truss, with a fabric cover. Then we will consider the same wing in which the fabric cover and the drag truss are replaced by a rigid cover—such as

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*After two years of flying he returned commercial work and passed into and one-half years in extensive technical research during which time he solved many difficult structural problems. Since joining the Fokker organization as chief structural engineer under Alfred J. Gurney, chief engineer, he has perfected new methods of structural calculation.*

*This article and several others have been prepared by Mr. Fradde in collaboration with Armand Thierlot of the Engineering Department, Fokker Aircraft Corporation.*

plywood or duralumin. We will use the study of the wing by indicating a very simple method for calculating any type of wing structure.

Then we will analyze other common indeterminate systems, such as the engine mount, landing structure,



(tension due to engine torque or tail load), tail surfaces and tail wheel.

## Study of Wing Structures

We will begin the analysis of wing structure by considering two considerations that will simplify the problem and on which the following methods are based. They will

be used in the calculation of wings considered as indeterminate structures. The first consideration refers to aerodynamics and the second one to the strength of materials.

They are:

- 1.—The Form of an airfoil section.
- 2.—The Elastic Center of a wing section.

1.—Form of an airfoil section and influence of the moment coefficient in the choice of an airfoil section.

THEOREM:

For every airfoil section, there exists a point for which the moment is independent of the angle of attack.

This point is called the "Focus" of the section (Fig. 1). The moment about the focus, constant with the speed, is nothing but  $LC_{m0}$ , due to the moment when the lift is 0 (i.e., being the chord) and  $C_{m0}$  the moment coefficient when  $C_L = 0$ .

Proof:—The air acting on the upper and lower surfaces of the wing may be represented by two forces, respectively  $R_u$  and  $R_d$  as shown in Fig. 2, which are functions of the angle of attack. Generally these two forces have not the same line of action and the resultant of all the air loads on the wing section may be replaced



by a force and a couple which will vary with the location of the point of application of the force, and with the angle of attack. We will demonstrate that there is a point for which the moment is constant.

Let us call  $k$  the position of the center of pressure, for a certain angle of attack  $i$ , expressed in per cent of the chord.

$$k = \frac{d}{c}$$

Where  $d$  is the distance from the center of pressure to the leading edge and  $c$  the chord.

The moment coefficient is given by the expression:

$$C_m = k(C_L \cos i + C_D \sin i)$$

The curve showing the variations of  $C_m$  as a function of  $C_L$  is generally very nearly a straight line (see Fig. 3). Therefore we may write,

$$C_m = C_{m0} + iC_L$$

and the moment is:

$$LC_m = LC_{m0} + iLC_L$$

It can be seen that the expression giving the moment contains two parts:  $LC_{m0}$ , which is constant and independent of the angle of attack, and  $iLC_L$ .

(1) Another expression for the point of interest of aerodynamic equilibrium must also be used in the calculations in the following operations.

It is evident that if we choose for the point of application of the force a point located at a distance from the leading edge equal to  $iC_L$ , the moment with respect to this point will be constant and equal to  $LC_{m0}$ .

A more rigorous proof of the above theorem, based upon higher aerodynamic considerations, would be outside of the scope of this paper.

The selection of an airfoil section is guided by the choice of certain coefficients that characterize the aerodynamic properties as well as the structural qualities of the airfoil considered.

Their relative importance depends upon the particular case considered; nevertheless, they are generally listed in the following order:

- a.—moment  $W/C_L$  (ft. sec.), which enables the plane to carry the maximum load per square foot at a given movement speed.
- b.—thickness at existing angle  $\left(\frac{t}{c}\right)$ , that requires the minimum thrust at this angle.
- c.—thickness ratio and number of centerlines—so important in the qualitative type of construction in giving to the span enough depth at the wing root, and to both tips a similar depth.

These three important considerations can be found in all aerodynamic data on wing sections.

But there is another very important factor that is generally not given with these listed above:

- d.—the moment coefficient at zero lift ( $C_{m0}$ ) which is connected with the travel of the center of pressure.

The usual curve of the position of the center of pressure against the angle of attack does not show well enough consequences of the displacement of the center of pressure on:

- 1.—the aerodynamic characteristics
- 2.—the stability
- 3.—the distribution of the loads on the wing structure.

But these can be easily found in determining the moment coefficient of the airfoil section at zero lift. The calculation of  $C_{m0}$  consists of planing:

$$C_{m0} = \frac{1}{c} (C_L \sin i + C_D \cos i)$$

and  $C_D$  against  $i$ , and reading  $C_{m0}$  for  $C_L = 0$ . The curve of  $C_{m0}$  is usually very nearly a straight line.

We will consider here the influence of  $C_{m0}$  only as far as its influence on the wing structure is involved, since our purpose is the analysis of the structure.

Generally the airfoil sections having a great center of pressure travel have a high  $C_{m0}$ . The torsional moment in the wing is evidently proportional to  $C_{m0}$ .

And according to the theorem demonstrated above, it is possible to determine the torsion in the wing in calculating the moment about the Focus of the airfoil section, or the resultant focus of the wing if different airfoil sections are used.

If the wing is designed in such a way that the center of elasticity, the derivative of which a given below coincides with the focus, the torsion of the wing is equal to the constant moment  $LC_{m0}$ ; if these two points do not

outside, the tension is equal to the above compressive moment minus the moment of the resultant passing through the focus about the elastic center.

The moment, being as shown in Fig. 4, increases the reaction on the rear spar and decreases the reaction on the front spar. [But this does not mean that it always causes to transform suddenly that moment into two vertical forces acting at the spars, because generally the direction of these reactions depends upon the entire wing structure and can only be determined with exactitude at every element along a part of the load is taken into account.]

If the airfoil section chosen has a small displacement of the center of pressure it has a small torsional moment, which means not only better aerodynamic characteristics



and inherent stability, but also lighter spars (especially the rear spar), lighter compression members or struts, whatever absorbs the tension in the wing structure.

#### 2.—Elastic Center of a Bifurcated

The spars, like the drag rods or rigid covering form a unit structure and therefore it is not correct to treat any of these elements separately.

It is dangerous to calculate each spar alone, and then to make assumptions to determine the loads in the compression members, unless these assumptions are checked by a test to destruction—which is very expensive for large wings.

The spars being generally not statically determined, can be solved only by the elasticity method.

In most cases, each spar is correctly analyzed and whenever there are more than two points of support the well-known equation of three moments is used or the



deflections calculated. These two methods are correct when they refer to the elasticity of the spar.

But the first one, the three moment equation, cannot be used with accuracy in the general case where the torsion varies along the span and when the neutral axis is not a straight line.

The mistake generally made in distributing the loads in the wing structure is to divide arbitrarily and comparing the stresses, as if each element were independent of the others, and neglecting the torsional moment etc.

The general positive method for analyzing a continuous wing structure is long and complicated.

The following method is not quite as rigorous, but is considerably shorter and easier.

For each wing section, a point may be determined such that a load applied at this point causes equal deflection in the two spars.

If a rigid covering is used, the wing structure may be considered as a box-wall box beam the walls being interconnected by the ribs, so that, if the ribs are stiff enough, each section returns to original form in torsion. A point may be determined such that a load applied at this point causes equal deflections of the two horizontal walls and equal deflections of the two vertical walls.

We will call this point the "Elastic Center" of the section.

If it is located at distances from the spars or walls is inversely proportional to their moments of inertia. As a matter of fact, for a certain wing section, the location of the elastic center depends upon the variation of the moments of inertia of the spars between the fuselage attachment and the section considered, as well as the length of the spars of the two spars have not the same length—but generally this may be neglected, and the elastic center determined separately for each wing section.

Any load on the wing structure may be replaced by a load acting through the resultant elastic center of that wing and a couple, or reaction moment. The couple, for instance, may be replaced by a load acting at the elastic center and a couple applied at the spars (which causes shearing stresses in the covering or a rigid covering is used), while the load which passes



through the elastic center is distributed between the two spars and causes a bending moment and a secondary torsional moment in each spar.

We will now show by a concrete example how the elastic center may be used in the analysis of any wing structure. We will first consider the case of the conventional type.

#### A.—Conventional Type

We will analyze a cantilever wing structure of a conventional type—that is a wing made of two spars, ribs, a drag strut, compression members and a fabric cover.

The purpose of this analysis is:

- 1.—To give a better method for distributing the aerodynamic loads on the spars.
- 2.—To give a method for calculating the compression members.

The usual way of analyzing a wing structure is based on two opposite assumptions:

- 1.—The resultant of the aerodynamic load is broken up into two parallel components applied at the spars, according to the location of the center of pressure, and regardless of the rest of the structure. Therefore the wing structure is assumed to be infinitely rigid.
- 2.—Then each spar is isolated and calculated as a beam; the three moment equation is used to determine the reactions at the deflections calculated, which means that the spar is no longer considered as being rigid, but is treated as an elastic system.

These two hypotheses are in contradiction.

In the method outlined below, the loads will not be distributed by regarding each spar as a separate unit, but according to the elastic properties of the wing structure considered as a whole. This is due to the fact that the presence of the transverse members changes the distribution of the external loads, because they transmit a part of the load from one spar to the other, leaving thus a relieving effect which is a function of their own elasticity. Thus it is seen that the drag rods in a wing have a two-fold task: first, to take care of the drag loads, and second to provide a certain amount of torsional rigidity in maintaining the original form of each cross section of



the wing by equalizing the deflections between the spars. And this important feature of the load spar structure should be taken into consideration in the distribution of the external loads.

#### Arbitrary Methods

The spars are treated separately. Thus it is no way of comparing the loads in the compression members. The Air Service and Department of Commerce require a margin of safety of 33 per cent and that the total load on the drag strut be assumed to act with an eccentricity equal to one-sixth of the spar depth.

#### Exact Method

The reactions at the fuselage attachment and at each compression member have to be determined.

The known and the assumed which define upon the reaction of the drag member  $A$  on the spar  $B$  (see Fig. 3) give us unknowns:

$X, Y, Z$  are the projections of the force on three axes of co-ordinates and  $M_x, M_y, M_z$  the three projections of the vector representing the moment.

The components  $X, Y, M_x, M_y$  having a reaction



influence on the vertical deflection of the spars, may be neglected.

Reaction for each drag member the reaction on the front spar are equal to the reaction on the rear spar.

Therefore we have only two unknowns per drag strut: a vertical force  $Z$  and a moment having its axis parallel with the rear axis  $M_y$ .

Thus, which are unknowns in the system upon drag members is  $2n + 2$ , where  $n$  is the number of drag members,  $n$  is the number of unknowns at the wing attachments of the spars. We have only six equations of equi-

librium. Therefore the system is generally not statically determinate.

The exact method consists of writing the equation of least work for the complete system not deriving this reaction with respect to each force and each moment in order to get the number of equations necessary to solve the system, that is  $2n + 2$  — 6. The equation is:

$$\Pi = \frac{1}{2} \int_0^L \left[ \frac{M_x^2}{EI} + \frac{M_y^2}{EI} + \frac{M_z^2}{EI} + \frac{P_x^2}{GA} + \frac{P_y^2}{GA} + \frac{P_z^2}{GA} \right] dx$$

#### Method Based on the Deflection

It is much easier to compute the deflection of each spar separately, it is then possible to determine, by means of the rule of "Reciprocity" of Maxwell, the load in the compression members due to the deflection of the deflection of the two spars cause these members cause the two spars to deflect together.

Just even this method generally gives long and tedious calculations.

#### Method Based on the Elastic Center

The consideration of the Elastic Center will simplify the problem very much. Let  $E$  be the Elastic Center defined as above. The distance from the Rear Spar to the Elastic Center is, with sufficient accuracy:

$$e = \frac{I_1}{I_1 + I_2} D$$

Where  $D$  is the distance between spars,  $I_1$  the moment of inertia of the front spar,  $I_2$  the moment of inertia of the rear spar.

Referring to Fig. 6:

Let  $P$  and  $P_1$  be the external loads acting on each spar (calculated by the approximate method, that is a sup-



posing the wing rigid and distributing the resultant around between the spars into two components inversely proportional to the distances to the center of pressure.) Let  $A$  be the distance from the front spar to the Elastic Center.

Under the loads  $P$  and  $P_1$  the front and rear spars will deflect and take positions  $A_1$  and  $A_2$ . (See Fig. 7).

If we replace these loads  $P$  and  $P_1$  by  $P$  and  $P_1$  and a moment  $M$  acting at the elastic center, the loads  $P$  will give to the spars equal deflection and therefore will not induce any load in the drag members which have only to transmit the moment  $M$ .

The force  $P$  applied at the elastic center may be broken up into two components, which will give two forces  $P'$  and  $P''$  acting respectively on the front and rear spars. The difference of the loads applied on the spar (that is, for the front spar, the difference between the load  $P$ , computed as outlined above, and the component  $P'$  of the force  $P$  which is really the load acting

on the spar) is the reaction of the drag member. We can make the calculations as follows:

Load really acting at the rear spar:

$$L_1 = (P + P_1) \frac{d}{D}$$

Load really acting at the front spar:

$$L = (P + P_1) - (P + P_1) \frac{d}{D}$$

$$L = \frac{(P + P_1)(D - d)}{D}$$

Drag member reaction (see Fig. B):

$$B = P_1 - (P + P_1) \frac{d}{D}$$

$$A = P - \frac{(P + P_1)(D - d)}{D}$$

or

$$B = \frac{P_1 - (D - d) - P d}{D}$$

$$A = \frac{P_1(d - D) + P d}{D}$$

$$A = -B$$

The reaction of these forces is equal to the moment acting at the elastic center.

Now, we have all the elements necessary to compute the stress in the drag member, which takes to bearing the resultant moment of the air reaction.

The load was distributed between the two spars by the normal method, but the calculation may also be made as follows:

Let  $L$  be the air reaction acting at the focus of the

airfoil section and  $M_1$  the moment,  $\alpha$  the distance between the focus and the elastic center, we have at the elastic center (see Fig. B):

$$\text{Force} = L K$$

$$\text{Moment} = (M_1 - L\alpha) K$$

(Because  $L$  is usually perpendicular to the chord direction  $K$  being the load factor.)



If  $D$  is the distance between the spars we have:

$$M = (M_1 - L\alpha) K$$

Reactions of Drag members on the spars (see Fig. B):

$$B = -A = \frac{(M_1 - L\alpha) K}{D}$$

True load on the front spar:

$$L \frac{(D - d) K}{D}$$

True load on the rear spar:

$$\frac{L K}{D}$$

It can be seen that these values are equal to the true loads on the spars determined above by the first method.

The second of these articles will appear in the July 20 issue of the *Aeronautical Engineering Section*.

## TECHNICAL PUBLICATIONS RECEIVED

*NACA Technical Note No. 385—Corrosion Induced by Durethane, VI. The Effect of Corrosion Accompanied by Stress on the Tensile Properties of Sheet Durethane*, by Henry S. Ranshaw, Bureau of Standards.

*NACA Technical Note No. 386—Curve Showing Ultimate Strength of Steel and Duralumin Tubing*, by Owen B. Ross.

*NACA Technical Memorandum No. 512—Contributions to the Technique of Landing Large Airships, Part I*, by G. Kraft, from *Zentralblatt für Flugtechnik und Motorluftschiffahrt*, September 26, 1938.

*NACA Technical Memorandum No. 513—Contributions to the Technique of Landing Large Airships, Part II*, by G. Kraft, from *Zentralblatt für Flugtechnik und Motorluftschiffahrt*, September 26, 1938.

*NACA Technical Memorandum No. 514—Technical Report of the 1938 Rhine Survey-Flight Contest*, by A. Lappin, from *Zentralblatt für Flugtechnik und Motorluftschiffahrt*, February 14, 1939.

*Air Corps Technical Report No. 2039—Determination of Structural Airplane Drag*.

*Air Corps Technical Report No. 2038—Dynamicometer Calibration Runs on Curtiss GV-250 Engines*.

*Air Corps Technical Report No. 2037, the Caudron*

*PT-3 (Airplane Wing Cable in Low and High Incidence Air Corps Technical Report No. 2040, Comparison of Wind Tunnel Tests with Flight Tests on a Number of Detachable Rotor Propeller Blade from the same Pilot*.

*Royal Aeronautical Society Lecture—Present Wind Tunnel Methods*, by Maurice A. Lippin, Director of the "Ruffel Laboratory", Aeronautical Engineer, Dr. of Science.

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*General Electric System—Aircraft Engine Problems*, by T. R. Rhin, Major and Aircraft Engineering Department, General Electric Company.

# Notes on the Design of Ailerons

By LIEUT. H. A. SUTTON

1. S. Army Air Corps

A STUDY of the control surfaces being used on many of the recent airplanes shows a general lack of appreciation of the importance of providing the best obtainable control characteristics. The continued use of inferior designs may be due partly to the necessity of depending primarily on empirical data and experience because of the absence of any satisfactory design theory.

Under such circumstances, it is important that the information gained from all sources be made readily available. Further use of a wind tunnel during the early stages of a design would correct many of the errors of judgment which occur, and it is to be expected that some use is not made of wind tunnels before large sums are spent in building new designs. A large proportion of the accidents now occurring could be prevented by improvement in controls and few things are more reassuring to the pilot in a tight situation than confidence in his ability to maintain control. Blaming the pilot for an expensive accident does not repair the airplane while even a slight improvement in control ability might have prevented the occurrence of such an accident.

## Control System Should be Simple

The design of aileron controls is particularly susceptible of improvement on many airplanes. Certainly nothing is more disappointing to a pilot than trying to operate a stiff and inefficient aileron control in rough air. Aileron controls should be simple in construction. A complicated system of pulleys and bell cranks may share elegance in looking interesting, but such systems are never satisfactory in operation because of excessive friction, wear in the numerous joints and general slackness which cause severe or serious control deficiencies which cannot be satisfactorily corrected. The right and left ailerons



A pilotcock steering the aileron on a Breda P-41 French plane.

should be connected independently to the operating control so that failure of one side will not cause a total loss of lateral control. Aileron control horns should be so located that excessive wing deflections will not be caused by aileron loads and the control system should be so arranged that it is not affected by deflections occurring in the airplane. Housing surfaces in all joints should be larger than is usual in present airplanes and should be easily lubricated.

The lateral control should be effective at low speed particularly during landing and take-off without being hard to operate at higher low or high speeds. This is generally hard to accomplish especially on large airplanes without the use of some type of balance. It is well known that aileron ratio is important in steering aerodynamic efficiency, and it is obvious that the aileron should be kept as far as possible from the longitudinal



Right and left sides of the aileron on a Thomas Motor observation airplane.

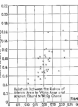
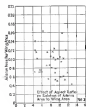
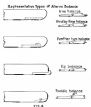
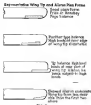
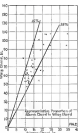
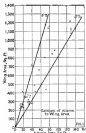
area. The aileron area required depends on these factors among others, but experience has shown that ailerons should have between 6 and 12 per cent of the wing area. For large airplanes whose lateral mass distribution is increased relatively by placing engines on the wings the ratio should be more nearly 10 than 6 per cent. This assumes that a good plan form is used as many of the old designs required more than this amount. Small airplanes with tapered wings require less aileron area, but very few airplanes have good lateral control with less than 6 per cent. Fig. 1 shows the properties in use as a number of representative airplanes.

The chord of an aileron should not be less than 15 per cent of the wing chord. A shorter aileron chord results in a loss of rolling moment obtainable for a given area and a very great increase in aileron chord increases the hinge moment and aileron jacking moments. The results plotted in Fig. 2 indicate that the normal limits are from 15 to 25 per cent. Aileron jacking moments can and should be eliminated, preferably by use of a differential motion of the aileron control surfaces to which the spread moving aileron

has a greater angular motion than the one which moves down. The angular motion should be at least 20 deg.

As the aileron span is increased toward the fuselage it becomes less efficient in providing rolling moment. The ratio of aileron span to distance between the longitudinal axis and the center of aileron area generally varies from 35 to 75, the smaller ratios being seen common on airplanes of large wing span and large aileron area. Since movement of the aileron has the effect of changing the lift over that portion of the wing of which it forms the trailing edge, it is significant that its span be kept large. Aileron aspect ratios vary between 5 and 11 with a general trend toward reduction in ratio of aileron area to wing area as the aspect ratio increases (see Fig. 3). Very long ailerons are liable to cause structural difficulties because deflection of the wing at aileron ends tends to lag behind.

The aileron plan form depends somewhat on the wing plan form. Load distribution tests have shown that rectangular or negatively raked wing tips are subject to high loading at the tips and it is obvious that ailerons should be kept out of such regions. Some typical plan



forms are shown in Fig. 5. Ailerons which extend beyond the wing tip are inefficient and cause high local hinge loads. Shrouded ailerons are not as satisfactory as those whose hinge axis lies approximately parallel to the lateral axis. In general ailerons should be approximately rectangular in plan form.

Aileron balances may be of many types, but in general the Free Balance is most satisfactory. In order to secure satisfactory control characteristics on an airplane the three control lines should be properly proportioned. If one control is twice as easy to operate as another the airplane will be difficult to fly properly. Satisfactory control operation may be secured without the use of a balance, but this condition is seldom found (especially on large airplanes) and use of the changes most frequently necessary in an airplane during its experimental development is a reduction of control force. Under usually so very satisfactory design conditions exist for determining the amount of balance desirable unless wing loaded tips are used. The aileron balance is not vary from 15 per cent to 30 per cent of the total aileron area depending somewhat on the type of balance used. Complete balance is of course not desired and the balance effect should be fairly uniform throughout the range of aileron movement and angle of attack so that the pilot

will retain a reasonably small positive "left" of the control. Fig. 6 illustrates the various types of aileron balance which have been generally used. Locked



Figure 7: Aileron as a Factor in Landing Phase

balances such as the Feather or tip type cause high local hinge loads and are less satisfactory than the Free or Handley Page types extending along the whole aileron span.

## TECHNICAL REVIEWS

*NACA Technical Report No. 370, Pressure Element of Constant Logarithmic Stiffness for Temperature-Compensated Aileron, by H. G. Brumberger and F. Condon*

The usual type of aileron control is a pressure element, the deflection of which are approximately proportional to pressure changes. An evenly divided aileron scale is secured by using a mechanism between the pressure element and the aileron which gives the required motion of the aileron. A temperature-compensated aileron was constructed at the Bureau of Standards for the Bureau of Aeronautics at the Navy Department which contained a manually operated device for controlling the multiplication of the mechanism to the extent necessary for temperature compensation. The introduction of this device made it difficult to adjust the multiplying mechanism to fit an evenly divided aileron scale. To meet this difficulty a pressure element was designed and constructed which gave deflection of the aileron proportional to altitude, that is, to the logarithm of the pressure. Mathematically, the logarithmic stiffness  $S$  of the element equals

$$S = \frac{A \log P}{dr}$$

from which is derived the deflection  $r$  for the change in pressure from  $P_1$  to  $P_2$

$$r = \frac{\log P_2 - \log P_1}{S}$$

The element consisted of a metal bellows of the "jellyfish" type supplied with an internal lateral spring which was designed so as to have a variable number of active coils. This report presents a description of said laboratory data relating to the special pressure element for the aileron. In addition equations which apply generally to springs and pressure elements of constant logarithmic

stiffness are developed, including the deflection and the spring between the coils in terms of the constants of the helical spring and pressure element.

*NACA Technical Report No. 389, Joint Report on Standardization Tests on NPL, RAE, and RAF, Model, by Walter S. Dool*

This report contains the wind-tunnel test data obtained in the United States on a 36 by 6 in. RAE 15 airfoil model prepared by the British Aeronautical Research Committee for international tests. Tests were made in co-operation with the National Advisory Committee for Aeronautics at the Bureau of Standards, Langley Memorial Aeronautical Laboratory, Massachusetts Institute of Technology, and McCook Field.

In addition to brief descriptions of the various wind tunnels and methods of testing, the report contains an analysis of the test data. It is shown that while in general the agreement is quite satisfactory there are two cases in which it is unsatisfactory. Since the lack of agreement in the latter is probably explained by errors known to be inherent in the methods of determining and applying corrections in these particular tests, it is concluded that the agreement obtained is more a matter of technique than a wind-tunnel characteristic.

*Development of WAFat Aircraft Construction, by S. C. Clark, Kansas City, Mo., and W. J. Gorton, New York, N. Y.*

In five pages the authors point out that steel tubing is coming into common use for fuselage construction and with this there is a trend toward greater use of welding. It is desirable to establish standard design for welded joints and in this and further research is needed. Suggestions are made for improvement in welding practice that will add to safety and secure uniformly well-made joints.

# Steel Aircraft Construction in Great Britain

By W. H. SAYERS

London & Paul Ltd.

OF THE materials commonly used in the construction of aircraft, steel ranks second only to spruce in the extent of its use. It is therefore somewhat remarkable that, as a rule when the airplane manufacturers of the three worlds are turning from timber to steel as their main structural material, only one reason has really seriously troubled the problem of using steel as a substitute for spruce.

The obstacle contained in the above paragraph is perhaps open to some misconstruction. In the first place it may be said that steel is now accepted almost everywhere as the normal material for linkage members, struts, and other parts, and secondly, that the diffi-

culty capacity for producing corrosion was so limited that the whole production was inadequate to the needs of the Airship constructors. The problem is required to steel was much more, possessing, and it was at first proposed to use solid drawn tubes of high tensile steel. Here difficulties of plate, as opposed to material applied sheet, insuperable but it was found impracticable to develop the production of grade improved surface on steel tubes of 40 or so tons ultimate strength which served in some part to alleviate the position. However it was speedily recognized that tubes of no available material would save the whole of the needs of the aircraft constructor and that for wing spars sections rolled in drawn to shape from thin strip steel presented a way out of the difficulties in question. Originally it was proposed to use a steel of the cold-rolled class with an ultimate tensile strength of about 40 tons per sq. in. and experimental tests of wing-spars of complete wings were produced and tested with satisfactory results as far back as 1915. In some very early examples of this class, however, alloy steels were used.

No analyses with steel wing structures were actually put into production during the War. The experimental work thus carried out proved that such structures could be built to compete in strength and weight with those of normal timber construction and also that the technical problem to be overcome before this type of structure could be adapted generally were of a very serious nature. At the same time it was felt that these difficulties were by no means insuperable and that the material advantages attending on the use of steel for this purpose, justified continued work along this line.

Encouraged by the Air Ministry, some two or three firms have steadily carried on this work of development ever since the end of the War with the result that today steel aircraft structures are the rule rather than the exception among new British Military airplanes.

## The General Problem of Steel Construction

The problem that has had to be solved in producing the steel airplane is that of making an ultra-light structure of adequate strength in an essentially dense material. Taking space as 28 lb. per cu. ft. with a resistance permissible stress of 5,000 lb. per sq. in., and weighing 490 lb. per cu. ft. steel density at least 85,000 lb. per cu. ft. to be able to compete on the strength-weight basis. There is no difficulty in obtaining steel which will give an ultimate tensile strength of over an elastic limit of over twice that figure, but this is by no means the same problem as developing the effective strength in members of the type required for airplane construction.

Where tension loads alone are concerned the specific tenacity of steel is sufficiently superior to that

of other materials to have caused no almost insuperable one, but for members taking end or bending loads, another set of considerations have to be taken into account.

For equal weight a steel member such as a hollow strut or bar-spar must have walls not more than one-sixteenth of the thickness of those of the corresponding spruce member. This means that the steel has to be used in the form of sheet a few thousandths of an inch thick. Any attempt to construct sections of the type used in normal engineering practice, of the usual dimensions necessary, is useless in members with such thin walls but by local deformation of the walls. This type failure is closely akin to the failure of a slender strut by buckling under a normal loading far smaller than that corresponding to the product of the cross-sectional area and the ultimate compressive strength of the material.

In very general terms, one may regard an airframe strip of such a thin-walled member as an elementary strut, having very little inherent stiffness of its own, but supported by the neighboring elementary strips. A series of such elementary strips forming a type of apple-cake web is then sufficient to provide the plane of its own width but not in that of its thickness.

If both the width and the length of this type are given compared to the thickness, however the type on edge will not serve as an effective beam. The edge is comparatively weak because it is the case that the control section is not very rigid in the longitudinal direction.

To overcome this type of failure, it is necessary to provide in any beam not only stiffness in the plane of the primary load which it is intended to resist, but also sufficient inherent stiffness in a plane at right angles to this, or some continuous lateral support. To give the necessary lateral stiffness for stability the edge under compressive load may be carried through a right angle, forming an angle section, or a flange strip may

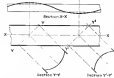


Fig. 3—Bending showing the diagonal wrinkles in the web caused by these stresses.

is added to form a "T" section. It will at once be recognized that types of section commonly used in beams for constant loads. The "T" channel used in the characteristic that the material is then it is arranged so as to provide not only stiffness in the plane of primary loading, but also an appreciable if lesser degree of stiffness in a plane at right angles to this primary plane.

It is usual to increase the distribution of material (web) in such sections to the degree to place an edge of it as possible at the greatest practicable distance from the neutral axis of the beam where it may most economically sustain bending. There is a large margin for such a statement, but quite apart from this, some web forms

as are, in fact, adapted would be necessary merely to guard against failure caused by lateral instability such as has been described.

Consider the case of the "T" section beam in Fig. 1 with flanges and webs of very thin material. Under loading the flange is compressed and it buckles at the edges and it assumes the wrinkled form shown by the dotted lines long before the applied bending moment reaches the value that the ordinary stress of beams would indicate as the failing loading. Complete failure of the beam will follow almost immediately after this wrinkling has set in.

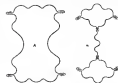


Fig. 4—The common types of corrugation of sections.

To prevent this type of failure it is necessary to stiffen the edges of the web, for example by flanging these edges as indicated in Fig. 2 (A).

If the ratio of unsupported width of flange to thickness of the web ( $W/T$  in Fig. 2 (A)) exceeds a certain limit the flange may still buckle between web and edge, the compression alternating between the two extremes shown dotted in Fig. 2 (A) and further intermediate deformations at B (Fig. 2) will be necessary. Such stiffening may equally be provided by adopting a longitudinally corrugated form such as shown at C (Fig. 2).

Having now satisfactorily defined the flange of the beam it will next be found that under loads the edge of the web will develop diagonal wrinkles buckling alternately in opposite directions as shown in Fig. 3 where VV and V'V' represent the planes of maximum distortion in the two directions. This form of distortion again leads to failure under loads much below those which would be indicated by ordinary methods of comparison. The remedy is again to give to the web increased longitudinal stiffness and similar methods to those adopted for the flange may be applied.

In the production of steel airplane wings from thin high tensile strip it is usual to give the necessary longitudinal stiffness by adopting a corrugated form as suggested in Fig. 2 (C).

The simple "T" section then shown is not however a satisfactory one owing to the difficulty of giving adequate stiffness to the webbing flange edges, and it is the almost invariable practice to interfere between successive corrugations, curved in the same sense others of opposite curvature.

Fig. 4 shows diagrammatically two types of construction fairly commonly adopted. An "A" is a box section formed from two corrugated webs and two corrugated flange strips laid together by four rows of rivets. An "H" is an "I" section whose flanges take the form of a closed tube. This is built up from seven strips and

des of using steel in any other form than that of tubes and for members which have to sustain other than loads are, so obvious that it is surprising that anyone should think of attempting it, when the light alloys such as duralumin, offer so much simpler means of achieving the desired result.

In Great Britain, however, the problems of producing steel structural members of all the types needed for the building of airplanes has been tackled with such success that today all British military aircraft, experimental types excepted, are of metal construction.

The majority of these machines are steel for all the main members, including wing spars, while the light alloys are in general relegated to a secondary role. This result has not been accompanied by any excessive structural weight; rather the reverse and it has relieved the Royal Air Force of most of the anxiety as to the effects of corrosion which are generally felt in regard to machines built from the aluminum alloys. The history of this particular development dates back to that period in the War when the difficulties of obtaining silver spruce of the quality and in the quantities required by the Allied Air Services were acute. The









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